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WIND-TUNNEL INVESTIGATION OF 20-PERCENT-CHORD  
PLAIN AND FRISE AILERONS ON AN NACA 23012 AIRFOIL

By F. M. Rogallo and Paul E. Purser

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WIND-TUNNEL INVESTIGATION OF 20-PERCENT-CHORD  
PLAIN AND FRISE AILERONS ON AN NACA 23012 AIRFOIL

By F. M. Rogallo and Paul E. Purser

SUMMARY

An investigation of several modifications of 20-percent-chord plain and Frise ailerons on an NACA 23012 airfoil was made in the NACA 7- by 10-foot wind tunnel. The static rolling, yawing, and hinge moments were determined and are herein presented for several angles of attack. The conditions under which aileron oscillation occurred were also determined.

The tests indicated that the oscillation of the Frise aileron was the result of an abrupt breakaway of the flow at the lower surface of the aileron nose when the aileron was deflected to some angle between  $-10^{\circ}$  and  $-20^{\circ}$ , the particular angle varying with the shape of the aileron and with the angle of attack of the airfoil. The flow breakaway was accompanied by a rapid increase in the hinge moment and, in general, by a decrease in the rolling moment. The tendency to oscillate was reduced or eliminated when a bulge or a nose slat was added to the lower surface of the aileron. The nose slat, moreover, increased the effective deflection range of the aileron.

The aileron-control characteristics were computed for a pursuit airplane with several of the aileron arrangements and with three assumed aileron linkages. The results presented illustrate the effects of variation of aileron floating tendency and of differential linkage and support the contention that proper adjustment of floating tendency by means of tabs, bulges, springs, or other devices, together with a suitable choice of differential linkage, offers a promising means of improving the control-force characteristics.

Internally balanced sealed ailerons with larger amounts of balance than the ailerons tested are considered promising.

## INTRODUCTION

The NACA has undertaken an extensive investigation of lateral-control devices for the purpose of developing new devices and of supplying more design data on devices previously developed.

A large amount of data has been published by the NACA on various arrangements of plain ailerons, but comparatively little has been published on Frise ailerons (references 1, 2, and 3). The greater part of the data available on Frise ailerons can be found in the Reports and Memoranda of the British Aeronautical Research Committee (references 4 to 9).

The investigation of this report was made primarily in an attempt to determine by means of wind-tunnel tests what modifications would be necessary to prevent the violent oscillations inherent in the Frise ailerons of a recently developed fighter airplane. The Frise ailerons tested were therefore designed to simulate the ailerons of a particular airplane. They are not representative of all Frise ailerons because, as stated in reference 3 and verified in the present investigation, the shape of Frise ailerons greatly affects their characteristics. The modifications made to the aileron during the investigation were, in general, modifications that could easily be made on the existing airplane. Tests of a plain sealed aileron without balance were included for comparison.

## APPARATUS AND METHODS

Tests were made in the NACA 7- by 10-foot closed-throat wind tunnel (reference 10) at an air speed of about 40 miles per hour, corresponding to a test Reynolds number of approximately 1,440,000. Some of the tests were repeated at an air speed of about 80 miles per hour, corresponding to a test Reynolds number of approximately 2,880,000. The test set-up is shown schematically in figure 1. The various 0.20c ailerons (fig. 2) were installed on the outboard 0.37  $b/2$  of the 4- by 8-foot NACA 23012 airfoil.

The airfoil was suspended horizontally in the wind tunnel with the inboard end attached to the tunnel wall to

simulate the semispan of a 16-foot wing. The attachment at the wall restrained the airfoil in pitch but not in roll or yaw. The forces necessary to restrain the outboard end of the airfoil were measured by the regular balance system. The rolling moments were computed from the difference in the vertical reactions at the outboard end with the aileron neutral and in the reactions with the aileron deflected; the yawing moments were similarly computed from the horizontal reactions. The lift coefficients of the airfoil in the tunnel were computed from the vertical outboard reaction with the aileron held at neutral and under the assumption that the lateral center of pressure of the semispan was  $0.45 b/2$  from the plane of symmetry.

The aileron was manually operated by a crank outside the tunnel near the inboard end of the wing, and the hinge moments were computed from the twist of a calibrated torque rod connecting the crank and the aileron. All the ailerons were approximately balanced statically and a relatively limber torque rod was used in order that any tendency of the ailerons to oscillate might be easily noticed. Because the capacity of the torque rod was necessarily limited, it was impossible to obtain all of the hinge moments in tests that were made at 80 miles per hour. When the hinge moments became too large for the capacity of the torque rod, the rolling and the yawing moments were determined with the aileron locked at the various deflections by means of a small clamp at the aileron.

## RESULTS AND DISCUSSION

### Coefficients

The results of the tests are presented in figures 3 to 10 as curves of rolling-, yawing-, and hinge-moment coefficients plotted against aileron deflection at several angles of attack for each aileron. The deflections at which the various ailerons began to oscillate are noted by arrows on the appropriate hinge-moment coefficient curves.

The symbols used in presenting the results are:

$C_L$  lift coefficient ( $L/qS$ )

$C_L'$  rolling-moment coefficient ( $L'/qbS$ )

$C_n'$	yawing-moment coefficient ( $N'/q_bS$ )
$C_h$	aileron hinge-moment coefficient ( $H_a/qS_a c_a$ )
$c$	wing chord
$c_a$	aileron chord measured along airfoil chord line from hinge axis of aileron to trailing edge of airfoil
$b$	twice span of semispan model
$S$	twice area of semispan model
$S_a$	aileron area behind hinge line
$L$	twice lift on semispan model
$L'$	rolling moment about wind axis
$N'$	yawing moment about wind axis
$H_a$	aileron hinge moment about hinge axis
$q$	dynamic pressure of air stream ( $\frac{1}{2} \rho V^2$ )
$\alpha$	angle of attack of airfoil in tunnel
$\delta_a$	aileron deflection, positive when trailing edge is down
$\delta_s$	nose slat deflection, positive when trailing edge is down
$G_{l,p}$	rate of change of rolling-moment coefficient $C_l'$ with helix angle $pb/2V$
$F_s$	stick force
$\theta_s$	stick angle
$R$	differential-crank length

A positive value of  $L'$  or  $C_l'$  corresponds to a decrease in lift on the model, and a positive value of  $N'$  or  $C_n'$  corresponds to an increase in drag on the model. Twice the actual lift, area, and span of the model were used in the reduction of the results because the model

represented half of a complete wing, as has been previously stated. No corrections have been made to the data for the effect of the tunnel walls. Although such corrections may be relatively large for this set-up, the data on the various modifications are comparable.

### Wind-Tunnel Data

Plain sealed aileron without balance.— The aerodynamic characteristics of the plain sealed aileron without balance are shown in figure 3. This aileron had fairly large hinge-moment-curve slopes ( $dC_h/d\delta_a$ ) and an upfloating tendency that increased with angle of attack. No oscillation of the aileron was noticed during the tests.

Plain aileron with  $0.326c_a$  balance.— The aerodynamic characteristics of the plain aileron with a  $0.326c_a$  symmetrical nose balance, sealed, unsealed, and with two arrangements of cover plates, are shown in figure 4. The characteristics of the unsealed aileron with only the top cover plate in place (fig. 4(a)) were very little different from those of the plain sealed aileron without balance except for the expected reduction in hinge-moment-curve slope. The same aileron with a sheet-rubber seal (fig. 4(b)) was more effective but had about the same hinge-moment characteristics as the unsealed aileron, probably because the seal was attached slightly behind the aileron nose.

The addition of the bottom cover plate to the airfoil with the balanced sealed aileron (fig. 4(c)) had comparatively little effect on the hinge-moment coefficients but produced an unexplained decrease in the effectiveness of the aileron. The only oscillation noticed in the tests of the plain balanced aileron was a slight oscillation at  $8^\circ$  angle of attack at an aileron deflection of  $-27.5^\circ$ . (See fig. 4(a).) Ailerons of this type but with larger amounts of balance are considered promising, and a systematic investigation of their characteristics is recommended.

Frise aileron with  $0.326c_a$  balance.— The aerodynamic characteristics of the Frise aileron with  $0.326c_a$  balance are shown in figure 5. The unsealed Frise aileron (fig. 5(a)) was less effective at a low angle of attack and slightly more effective at a high angle of attack than the unsealed plain aileron (fig. 4(a)). The Frise aileron had

an upfloating tendency and a very small hinge-moment-curve slope at low deflections, but at high deflections ( $10^\circ$  and  $-20^\circ$ ) the differences in hinge-moment coefficients were as large as those of the plain ailerons. The small hinge-moment-curve slopes at low deflections may be a contributing factor to control-free lateral instability.

Comparison of the results of figures 5(a) and 5(b) shows the scale effect on the characteristics of the Frise aileron. The increased speed increased the effectiveness of the aileron at all angles of attack.

The addition of a sheet-rubber seal at the nose of the Frise aileron (fig. 5(c)) increased the rolling-moment effectiveness of the aileron. The location of the seal (at aileron nose instead of on upper surface near slot lip) decreased the effectiveness of the balance, probably because the seal prevented the pressures on the wing lower surface and ahead of the aileron from acting on top of the aileron nose. The seal also changed the upfloating tendency to a downfloating tendency. It is thought that a seal near the upper surface of the airfoil would increase the rolling moment without reducing the effective balance.

The addition of a trailing-edge tab, deflected  $-15^\circ$ , to the Frise aileron with  $0.326c_a$  balance (fig. 5(d)) had somewhat the same effect on the characteristics of the aileron as did the addition of the seal, partly because of the increased size of the aileron. This increase in size was not considered in the computation of the hinge-moment coefficients. The aileron with the tab, however, was not quite so effective as the aileron with the seal.

Neither the increase in speed nor the addition of the seal or tab had much effect on the oscillatory tendencies of the Frise aileron with  $0.326c_a$  balance. This aileron oscillated rather violently at deflections ranging from  $-16^\circ$  to  $-25^\circ$ , depending on the angle of attack. It is apparent from these data and from unpublished results of flight tests of two different installations of Frise ailerons that the presence of oscillation, and the aileron deflection at which it occurs, is dependent on the particular installation (shape, surface finish, rigidity of the system, etc.). In some installations, Frise ailerons deflected upward nearly  $20^\circ$  have shown no apparent tendency to oscillate. It is not generally considered advisable, however, to permit such large deflections for this type of aileron.

A study of the data and an observation of tufts located on the lower surface of the aileron made it apparent that the oscillation of the aileron was not what is generally called aileron flutter. With the airfoil at an angle of attack of  $8^\circ$  the aileron was deflected to  $-17^\circ$  before the flow began to break away from the aileron lower surface; below this angle ( $-17^\circ$ ) the hinge moments were small. At deflections of  $-17^\circ$  to  $-20^\circ$  the hinge moment increased rapidly and at  $-20^\circ$  the flow had completely broken away from the lower surface of the aileron. The elasticity of the torque rod allowed the large hinge moment occurring at  $\delta_a = -20^\circ$  to return the aileron to a deflection of  $-15^\circ$  where the flow became smooth and the hinge moment became small; here the spring effect of the torque rod again deflected the aileron to  $-20^\circ$  and once again the flow broke away and the large hinge moment decreased the aileron deflection. This process continued until the aileron was moved to a different angle by the crank. Both portions of the hinge-moment-coefficient curve had stable slopes and, since the tufts showed that the break in the curve was a stalling phenomenon, the actual variation of the hinge-moment coefficient during the oscillation is probably that indicated by the arrows in figure 6.

The Frise aileron with  $0.326c_a$  balance was then equipped with a  $0.01c$  lower-surface bulge (fig. 7) to prevent flow separation at the aileron nose by increasing the radius of curvature. The same purpose could probably have been accomplished by cutting away part of the original aileron nose. The bulge slightly increased the rolling-moment effectiveness of the aileron and decreased the hinge-moment-curve slope at high deflections but produced an unstable hinge-moment-curve slope in part of the negative deflection range. The bulge also caused the upfloating tendency of the aileron to change to a downfloating tendency at low angles of attack. This change in floating tendency will tend to increase the stick forces when a conventional differential system is used and could probably be counteracted by an additional bulge on the upper surface of the aileron near the trailing edge or by a forward movement of the point of maximum thickness of the lower surface bulge.

No oscillatory tendencies were noticed in the tests of the aileron with the bulge.

Frise aileron with  $0.278c_a$  balance.— The aerodynamic characteristics of the Frise aileron with  $0.278c_a$  balance

are shown in figure 8. This aileron (fig. 8(a)) had a greater hinge-moment-curve slope than the Frise aileron with  $0.326c_a$  balance, as was expected. The change in the amount of balance had little effect on the oscillatory tendencies of the aileron.

The addition of the  $0.01c$  lower-surface bulge (fig. 8(b)) had approximately the same effect on the Frise aileron with  $0.278c_a$  balance as it had on the Frise aileron with the larger balance. The bulge gave the aileron a downfloating tendency, decreased the hinge-moment-curve slope, and apparently eliminated the oscillatory tendencies of the aileron. The addition of a sheet-rubber seal at the nose of the aileron with the bulge (fig. 8(c)) slightly increased the rolling-moment effectiveness of the aileron but altered the hinge-moment characteristics surprisingly little relative to the large effect shown in figure 5. The seal did not change the oscillatory tendencies of the aileron.

Frise aileron with  $0.293c_a$  balance.— Two tests (fig. 9) were made of a Frise aileron with a  $0.298c_a$  balance of a square, unconventional shape. The upper surface of the nose remained within the wing contour at deflections up to  $-20^\circ$ . It was thought that this modification might decrease the oscillatory tendencies of the aileron. Instead, the aileron was less effective, had larger hinge-moment coefficients than the aileron with the  $0.278c_a$  balance, and still oscillated at  $\delta_a = -13.5^\circ$ . (See fig. 9.)

Frise aileron with  $0.278c_a$  balance and a nose slat.— The aerodynamic characteristics of the Frise aileron with  $0.278c_a$  balance and a nose slat (NACA 22 section) are shown in figure 10. The nose slat set at  $17^\circ$  (fig. 10(a)) increased the rolling-moment effectiveness and balance and reduced the oscillatory tendencies of the aileron. Increasing the slat angle to  $28^\circ$  (fig. 10(b)) made the aileron almost as effective as the plain sealed aileron without balance, reduced the hinge-moment coefficients at high deflections, and improved the oscillatory tendencies still more. Adding a sheet-rubber seal at the nose of the aileron with the slat at  $28^\circ$  (fig. 10(c)) had little effect on the characteristics of the aileron except to increase the rolling-moment effectiveness slightly at moderate deflections.

It should be noted that the slat span was only  $0.31 b/2$

while the aileron span was  $0.37 b/2$ ; a slat the full length of the aileron would probably be more effective. Also, since only two slat arrangements were tested, it is probable that neither the deflection nor the position of the slat was the optimum. These tests indicated, however, that slats may be very useful on control surfaces.

#### Application of Data

The aileron-control characteristics of a pursuit airplane (fig. 11) equipped with several aileron arrangements with an equal up-and-down linkage ( $+15^\circ$ ) and a differential linkage of the same total deflection (fig. 12) have been computed and are presented in figures 13 and 14. For simplicity, these lateral-control characteristics were computed from the data in figures 3, 4(b), 5(a), 5(c), 8(a), 8(b), 10(b), and 10(c) (the uncorrected aerodynamic characteristics of the ailerons) without taking account of the difference in wing plan form. The effects of rolling, moreover, have not been considered; these effects will be discussed later. Because the assumptions and the methods of computation followed herein are the same as those in references 11, 12, and 13, the computed characteristics of the several reports are thought to be comparable. The lift coefficient of the airplane at any particular angle of attack was assumed to be that of the airfoil in the wind tunnel, this coefficient being computed as previously described under Apparatus and Methods.

As was expected, the data in figures 13 and 14 show that the Frise ailerons, in general, had smaller stick forces than plain ailerons of the same size for a given value of rolling-moment coefficient. The plain ailerons, however, were more effective than the Frise ailerons at the same deflections. The adverse (negative) yawing-moment coefficients of the two types of aileron were about the same except near full aileron deflection, where the Frise ailerons had better yawing-moment characteristics but also had high stick forces.

With an equal up-and-down deflection the 0.15c plain sealed aileron with  $0.35c_n$  balance of reference 11 had approximately the same effectiveness and yawing-moment characteristics as the 0.20c Frise unsealed aileron with  $0.326c_n$  balance and had only slightly larger stick forces. The plain aileron also has in its favor the fact that the amount of aerodynamic balance could probably be increased

enough to give it lower stick forces than those of the Frise aileron at full deflection without overbalancing in the low-deflection range. The fact that the aileron would not be overbalanced in the low-deflection range would reduce the possibility of the occurrence of control-free lateral instability. The 0.15c plain aileron, moreover, should not tend to oscillate.

At low speeds the conventional differential system gave lower stick forces than the equal-deflection system. At high speeds, however, except at small deflections, the two systems gave about equal stick forces. (See figs. 13 and 14.)

During the analysis of the data it became apparent that the use of a reversed differential (down aileron deflected more than up aileron) might be advantageous when the aileron had a downfloating tendency. This possibility may be inferred from the analysis of aileron-linkage systems presented in reference 14. The differential shown in figure 12 was reversed and applied to a Frise aileron with and without a seal (fig. 15). The use of the reversed differential slightly increased the adverse (negative) yawing-moment coefficients. (See figs. 13(b), 14(b), and 15.)

Figure 16 is a comparison of the stick forces of an airplane equipped with the sealed Frise aileron with three linkages: conventional differential, equal up-and-down, and reversed differential. The combination of downfloating tendency and reversed differential gave a considerable reduction in the high-speed stick forces. An increase in angle of attack decreased the downfloating tendency of the aileron and it was estimated from other data that the sealed aileron would float up slightly at  $\alpha = 15^\circ$ . On the basis of this estimation an approximate curve was drawn in figure 5(c) and from this curve the low-speed stick forces (fig. 16) were computed. At low speed the reversed differential and the upfloating tendency increased the stick forces. This increase in stick force would give more feel to the stick at low speed and less variation of stick force with speed.

Because the floating tendency of ailerons may be controlled by the use of springs, tabs, or bulges and by nose treatment, adjustment of floating tendency and of differential offers a promising means of controlling stick forces and stick-force variation with speed.

The effects of rolling have not been considered in the computed characteristics of figures 13 to 16. The characteristics presented are those that would exist if the airplane were restrained in roll and yaw, as is generally true of a wind-tunnel model. An airplane actually begins to roll almost immediately after the ailerons are deflected. In order to illustrate the effect of rolling upon the stick force required to produce a given rolling-moment coefficient, comparative curves are given in figure 17. The solid lines represent the static condition, in which the airplane is not permitted to roll. The broken curves represent the condition in which the airplane is rolling with a velocity such that the rolling moment due to roll is numerically equal to the rolling moment due to aileron deflection. (See reference 15 or 16.) The value of  $C_{l_p}$  for the illustrative airplane was estimated as 0.45 from the curves of reference 15 or 16.

The differences between the curves shown by the solid and the broken lines of figure 17 are almost entirely the result of the variation of aileron hinge moment with angle of attack. If comparative curves similar to those of figure 17 were drawn for plug-type ailerons (see reference 13), a reduction of stick force at a given rolling-moment coefficient would also be shown, but for these ailerons the reduction would be primarily the result of the increase of rolling-moment coefficient with angle of attack.

## CONCLUSIONS

The oscillatory tendency found in some flight installations of Frise ailerons was shown by the wind-tunnel tests to be the result of an abrupt breakaway of the flow at the lower surface of the aileron nose when the aileron was deflected. This breakaway occurred at an aileron deflection between  $-10^\circ$  and  $-20^\circ$ , the aileron deflection varying with the angle of attack of the airfoil and with the shape of the aileron.

When the flow breakaway occurred, the hinge moment increased rapidly and the rolling moment usually decreased. It appears that Frise ailerons should be so designed that they will not be deflected to the angle at which breakaway occurs. The useful range of Frise ailerons may sometimes be increased by the addition of a nose slat or a bulge on the aileron lower surface.

Internally balanced sealed ailerons with larger amounts of balance than those tested are considered promising, and a systematic investigation of their characteristics is recommended.

Because the floating tendency of ailerons may be controlled by the use of springs, tabs, or bulges and by nose treatment, adjustment of floating tendency and of differential offers a promising means of controlling stick forces as well as the variation of stick force with speed.

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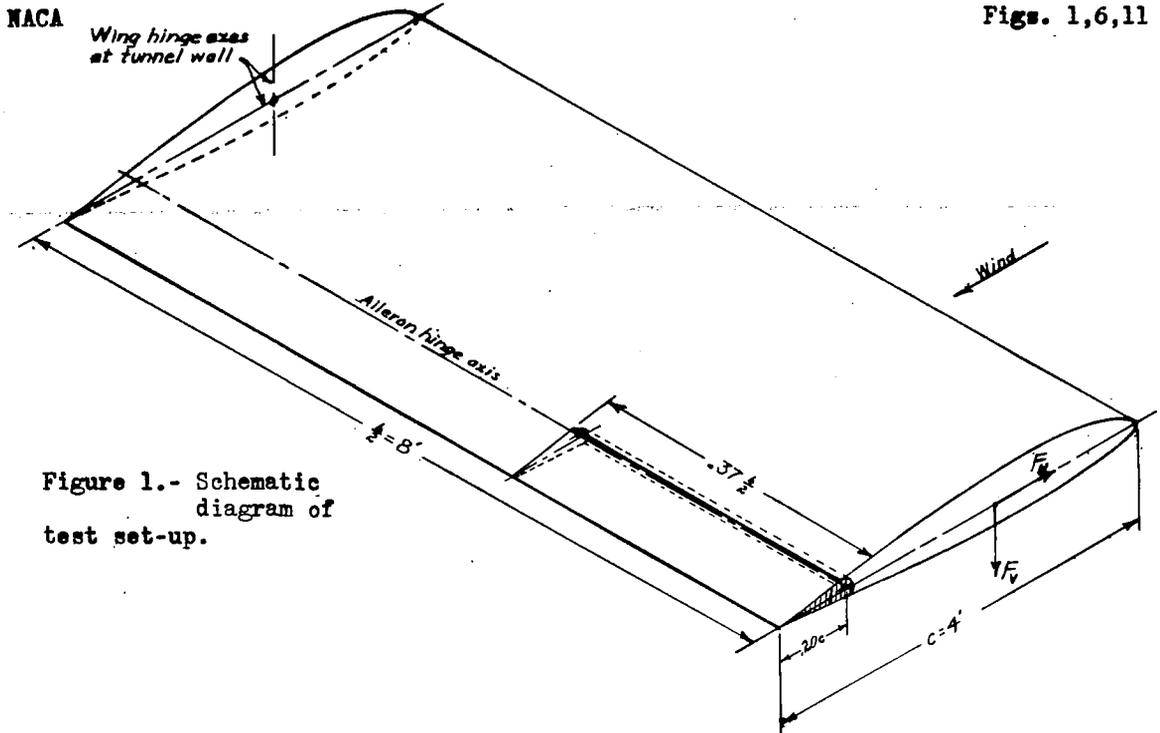


Figure 1.- Schematic diagram of test set-up.

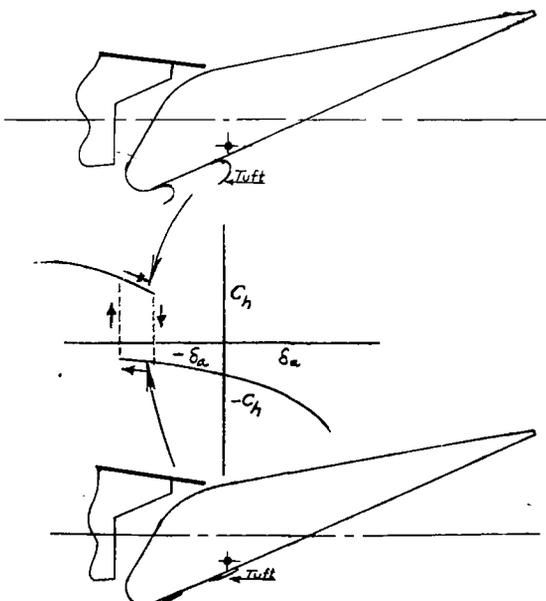


Figure 6.- Tuft action on the lower surface of an up-deflected Frise aileron, showing cause of oscillation.

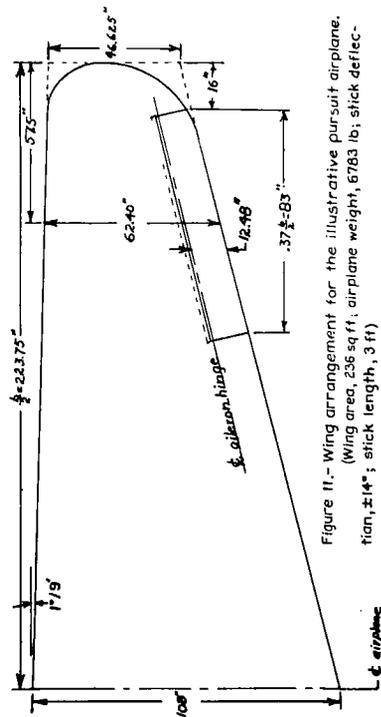


Figure 11.- Wing arrangement for the illustrative pursuit airplane. (Wing area, 256 sq ft; airplane weight, 6783 lb; stick deflection,  $\pm 14^\circ$ ; stick length, 3 ft)

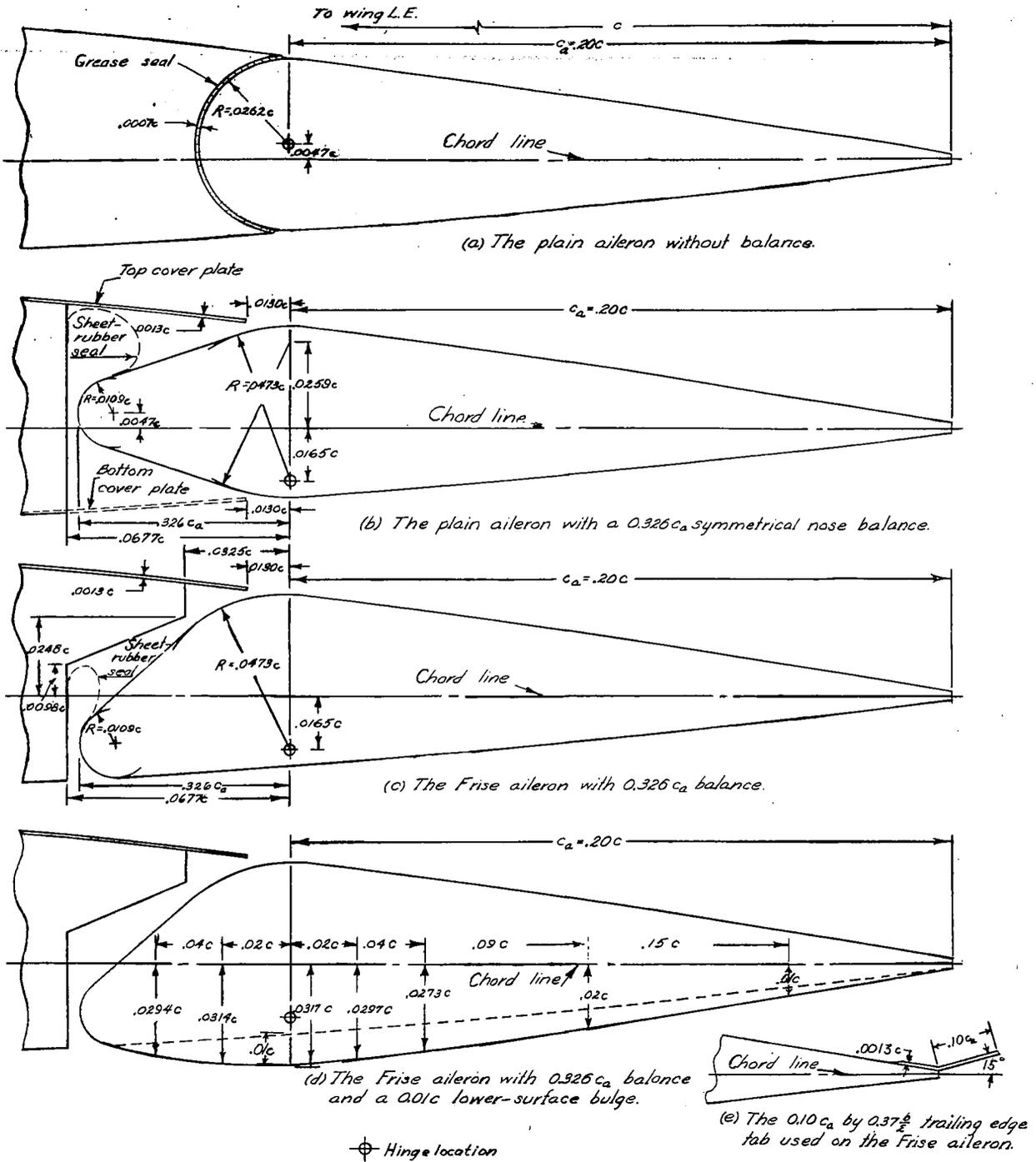


FIGURE 2.- The  $0.20c$  by  $0.37\frac{1}{2}$  ailerons tested on the 8-foot semispan NACA 23012 airfoil.

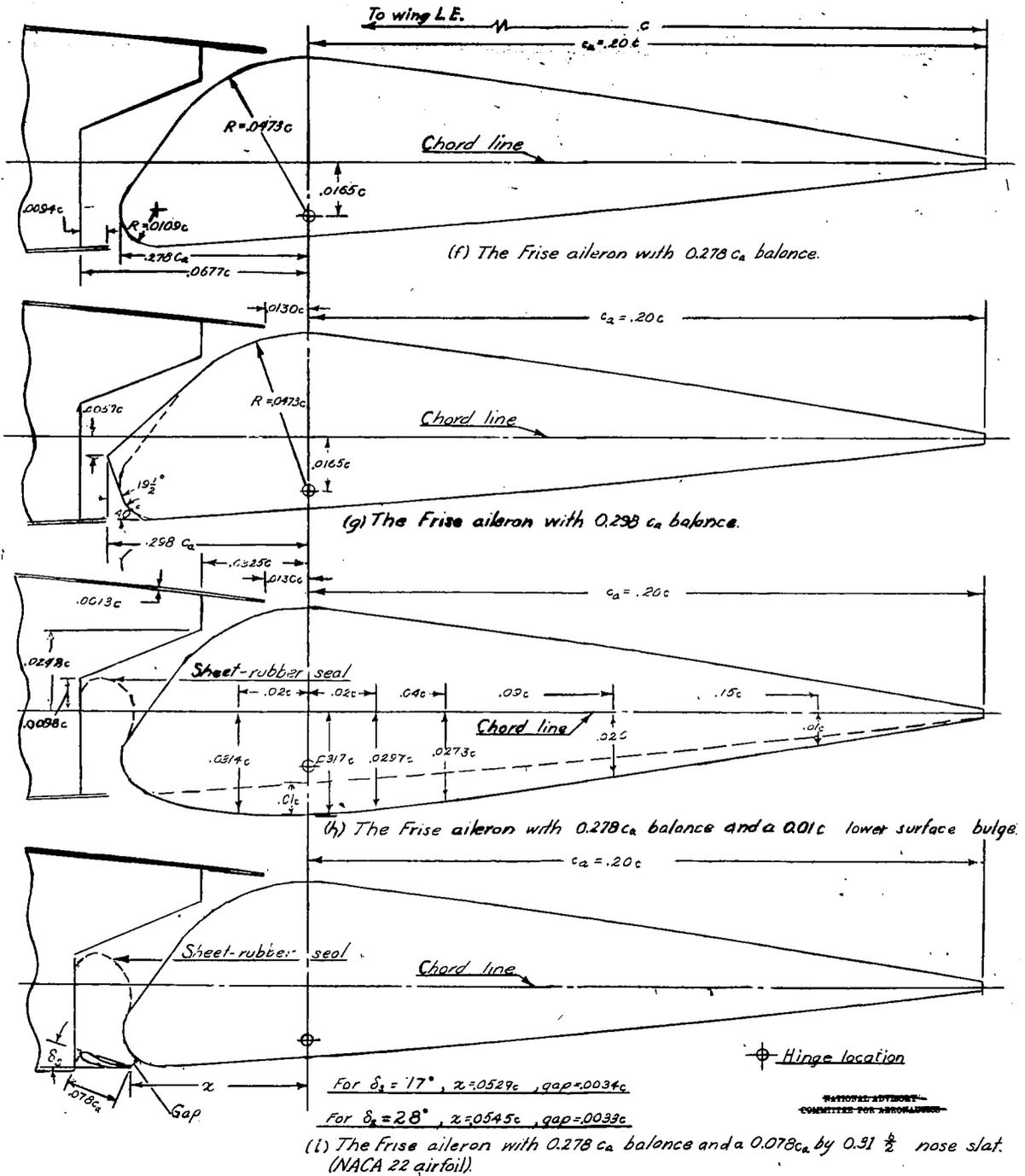


Figure 2- Concluded.

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Figs. 3, 4a

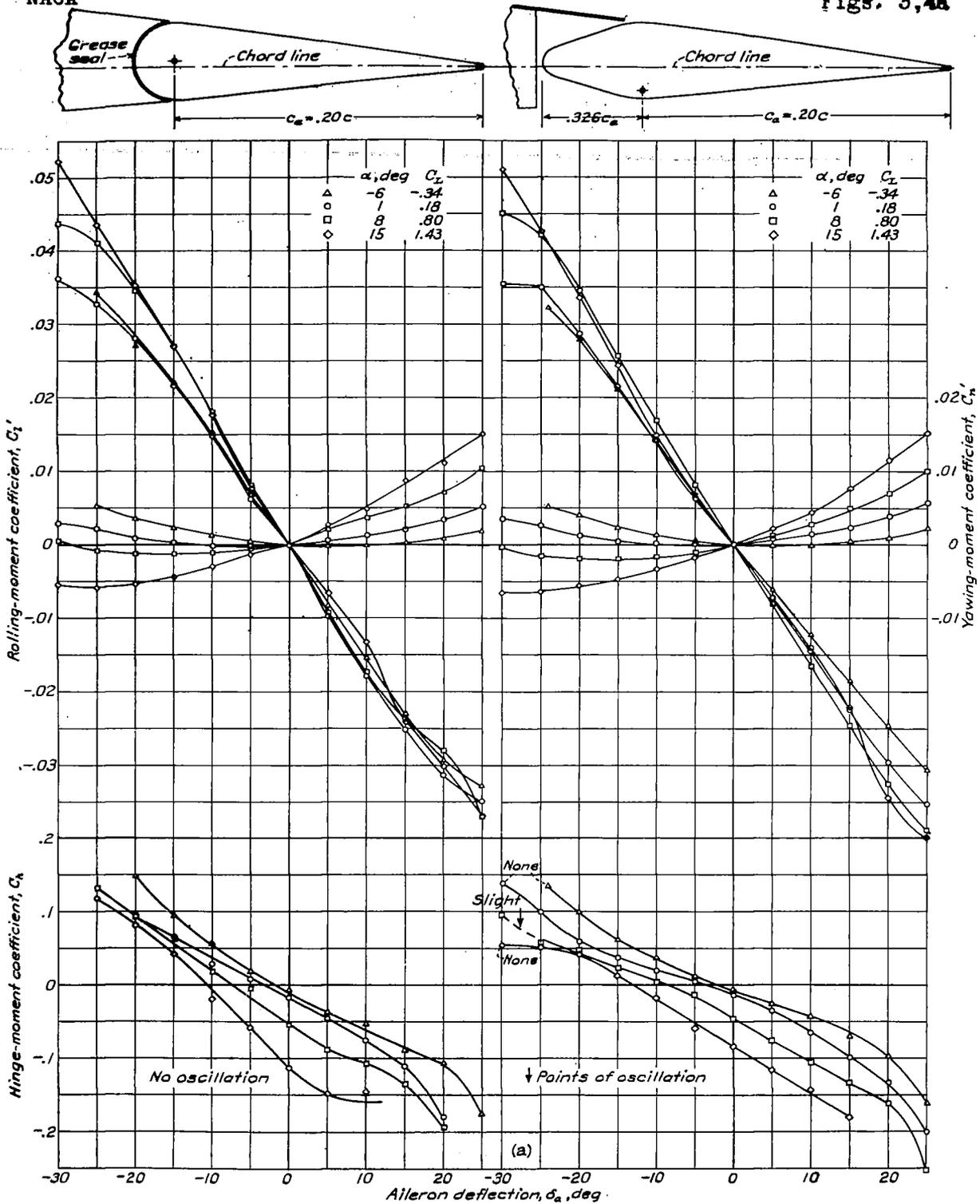
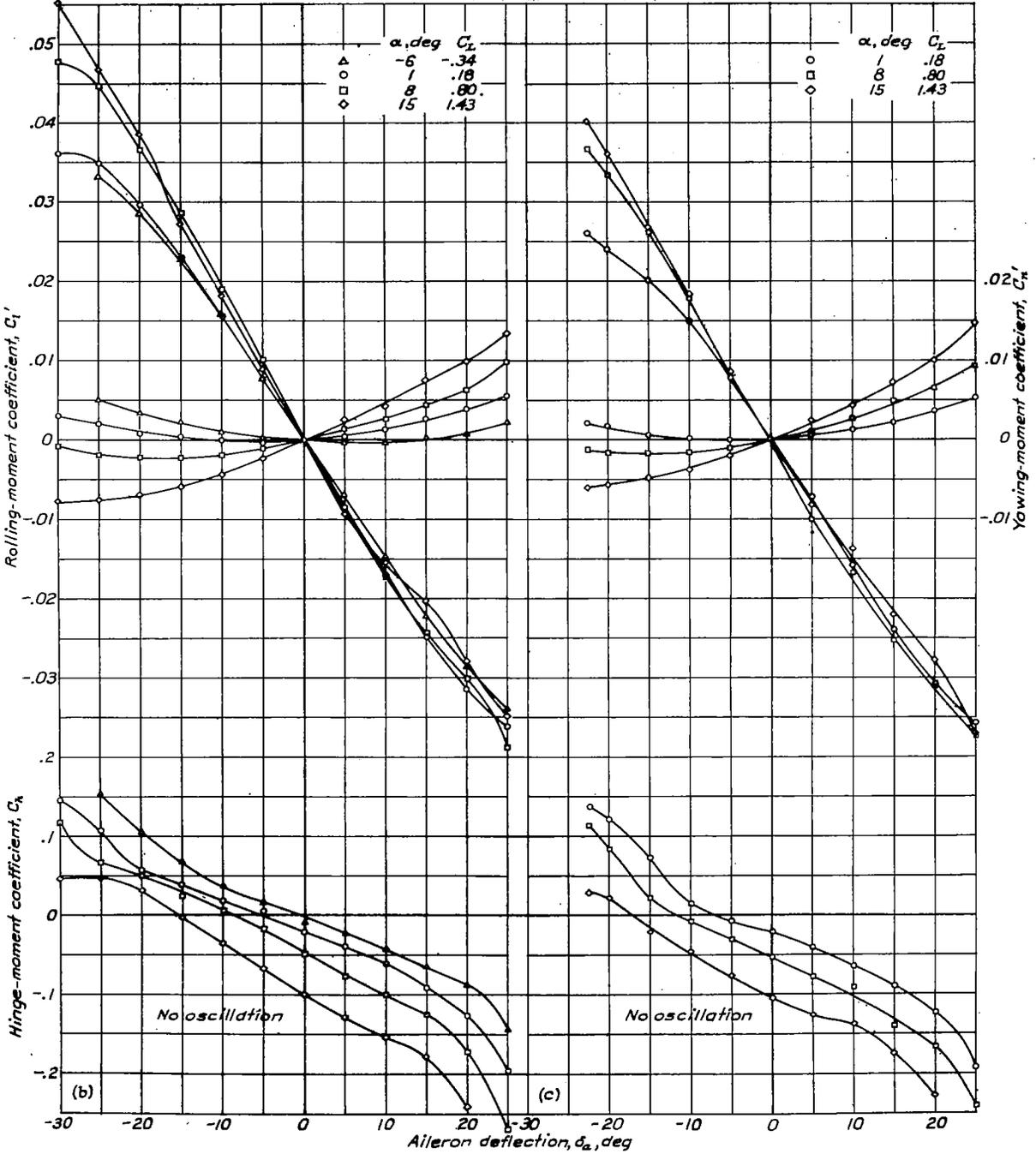
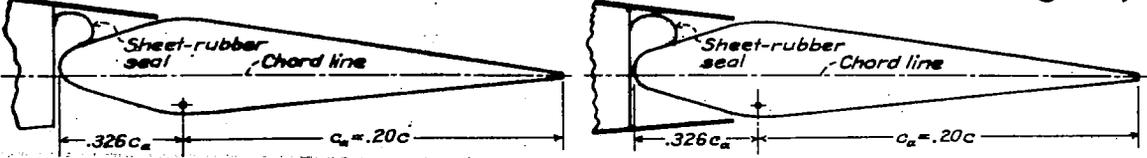


Figure 3.- Aerodynamic characteristics of a 0.20c by 0.37 b/2 plain grease-sealed aileron without balance on an NACA 23012 airfoil. V, 40 mph.

Figure 4a to c.- Aerodynamic characteristics of a 0.20c by 0.37 b/2 plain aileron with a 0.326c<sub>a</sub> symmetrical nose balance on an NACA 23012 airfoil. V, 40 mph.

Fig. 4b,c

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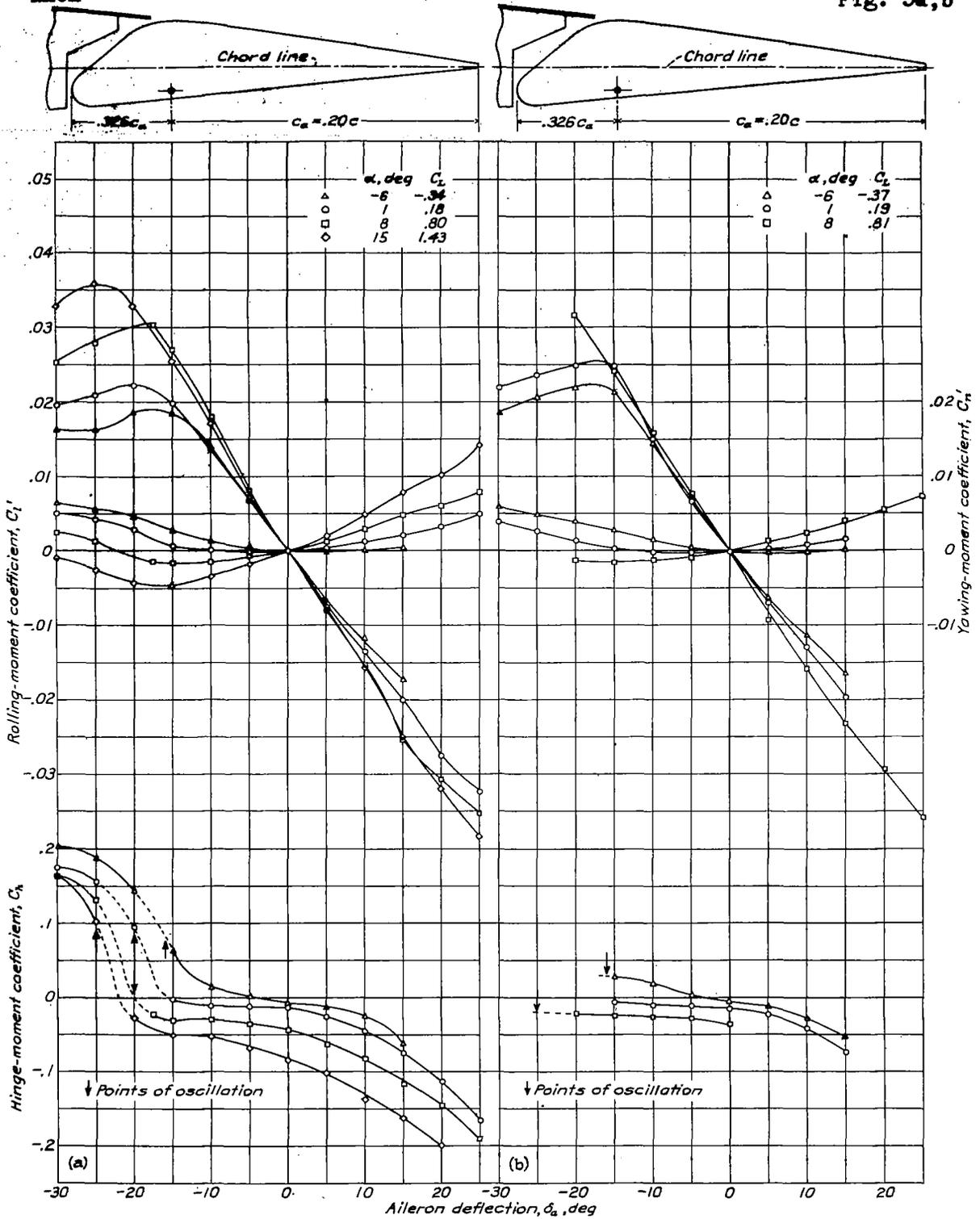


(b) Sheet-rubber seal; top cover plate only. (c) Sheet-rubber seal; both cover plates.

Figure 4b,c.

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Fig. 5a,b



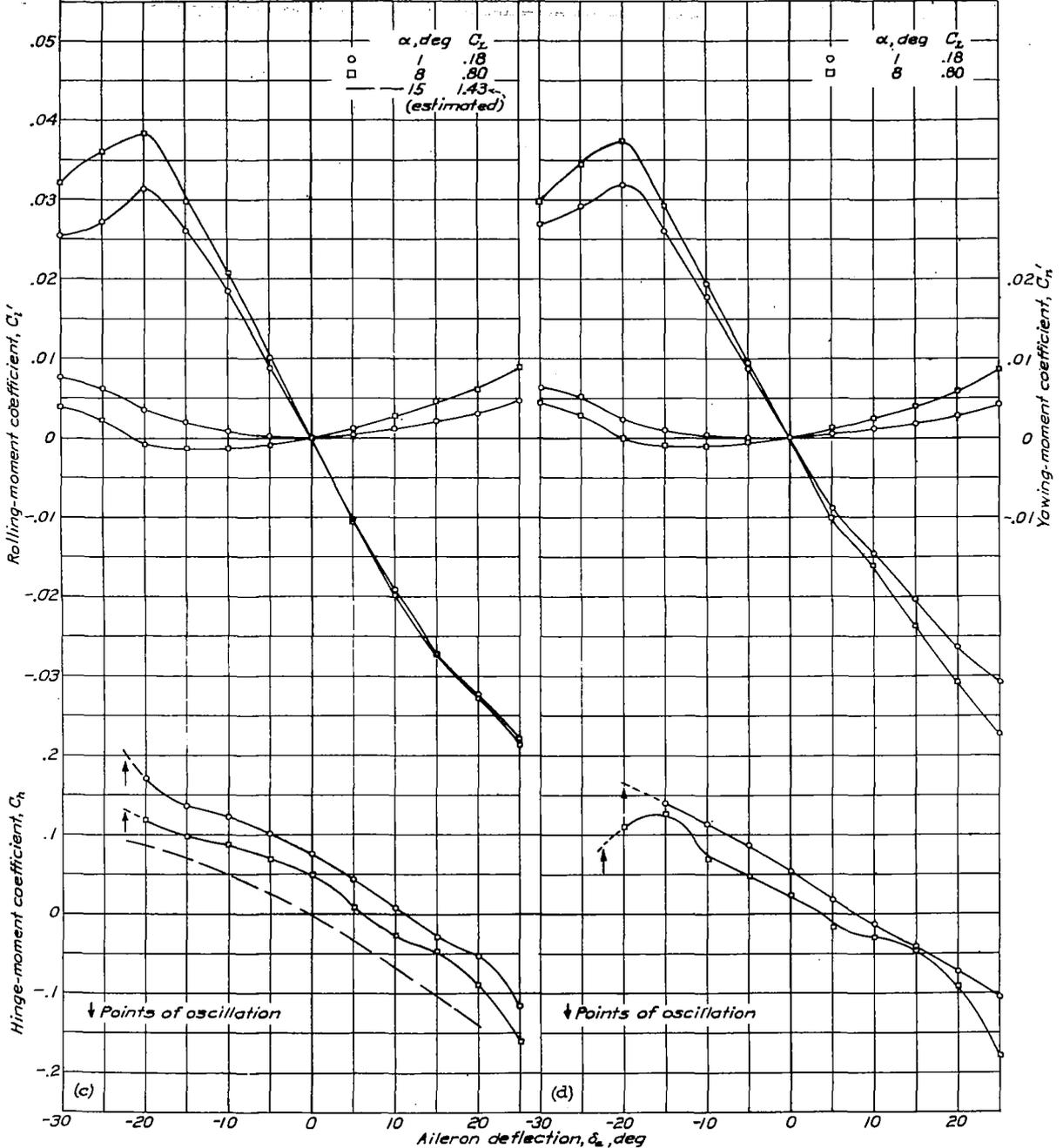
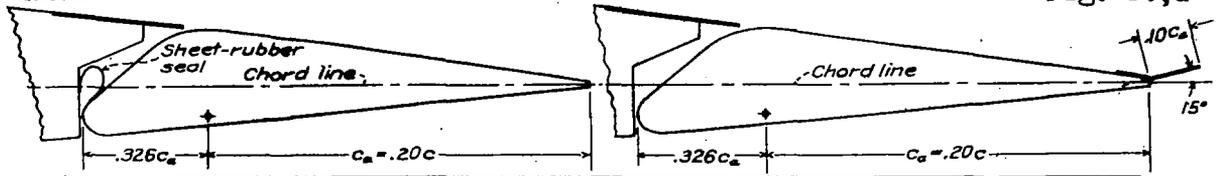
(a) No seal;  $V$ , 40 mph.

(b) No seal;  $V$ , 80 mph.

Figure 5a to d.- Aerodynamic characteristics of a 0.20c by 0.37 b/2 Frise aileron with 0.326c<sub>a</sub> balance on an NACA 23012 airfoil.

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Fig. 5c,d



(c) Sheet-rubber seal; V, 40 mph. (d) No seal; V, 40 mph;  $0.10c_a$  by  $0.37 b/2$  tab;  $\delta_t, -15^\circ$ .

Figure 5c,d.

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Figs. 7, 8a

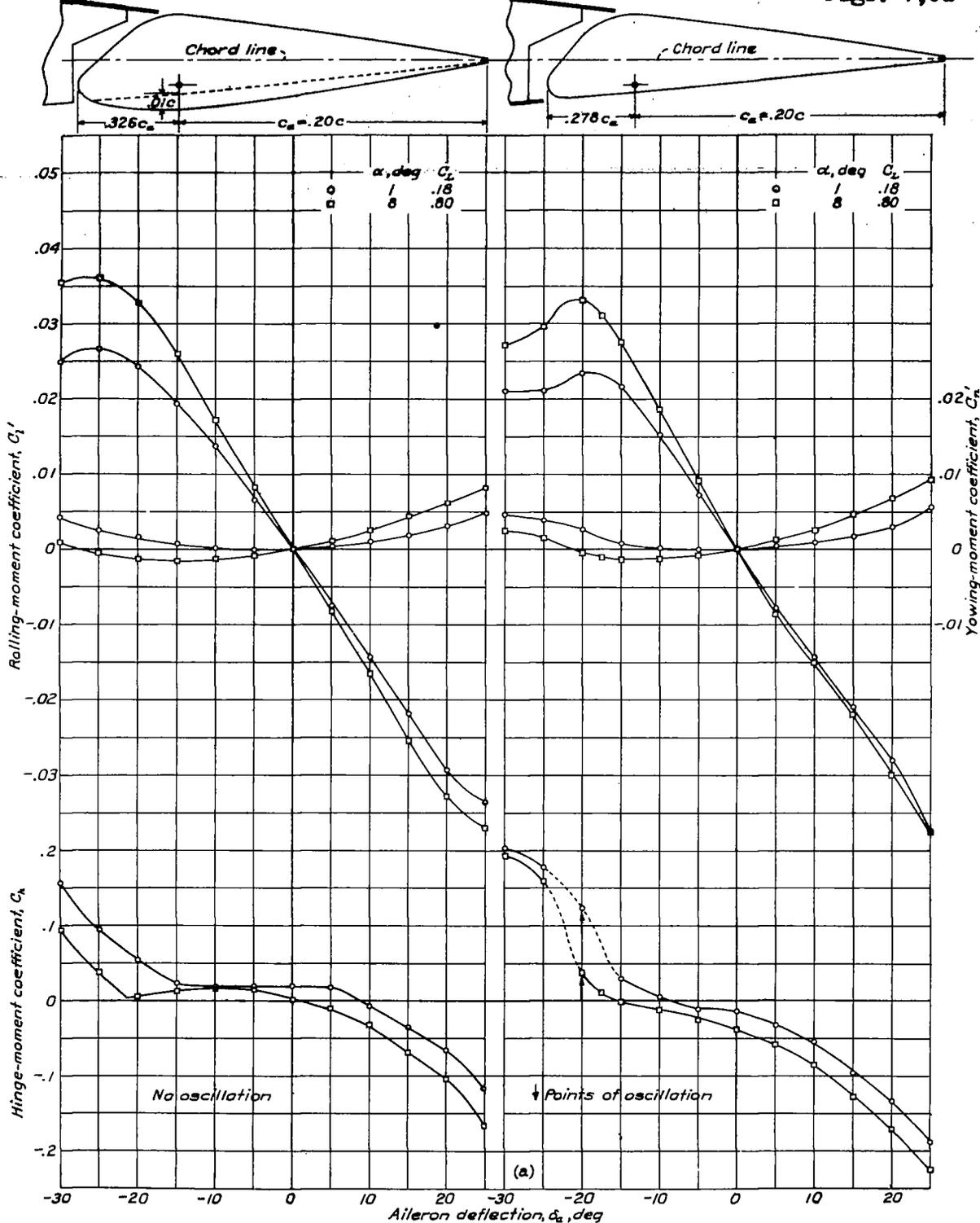
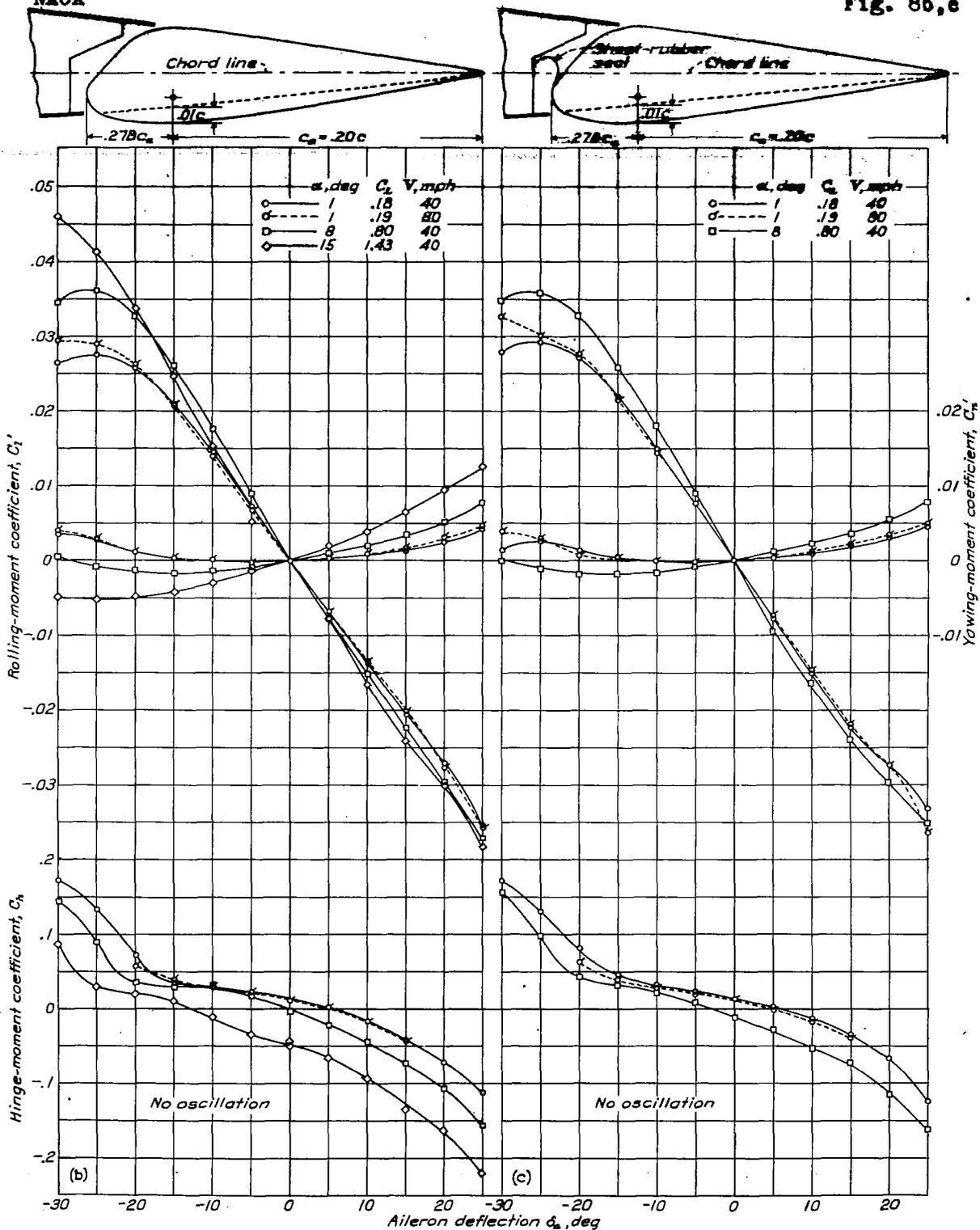


Figure 7.- Aerodynamic characteristics of a 0.20c by 0.37 b/2 Frise aileron with 0.326c<sub>a</sub> balance and a 0.01c lower-surface bulge on an NACA 23012 airfoil. No seal; V, 40 mph.

(a) No seal; V, 40 mph.  
Figure 8a to c.- Aerodynamic characteristics of a 0.20c by 0.37 b/2 Frise aileron with 0.278c<sub>a</sub> balance on an NACA 23012 airfoil.

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Fig. 8b,c



(b) No seal; 0.01c lower-surface bulge. (c) Sheet-rubber seal; 0.01c lower-surface bulge.

Figure 8b,c.

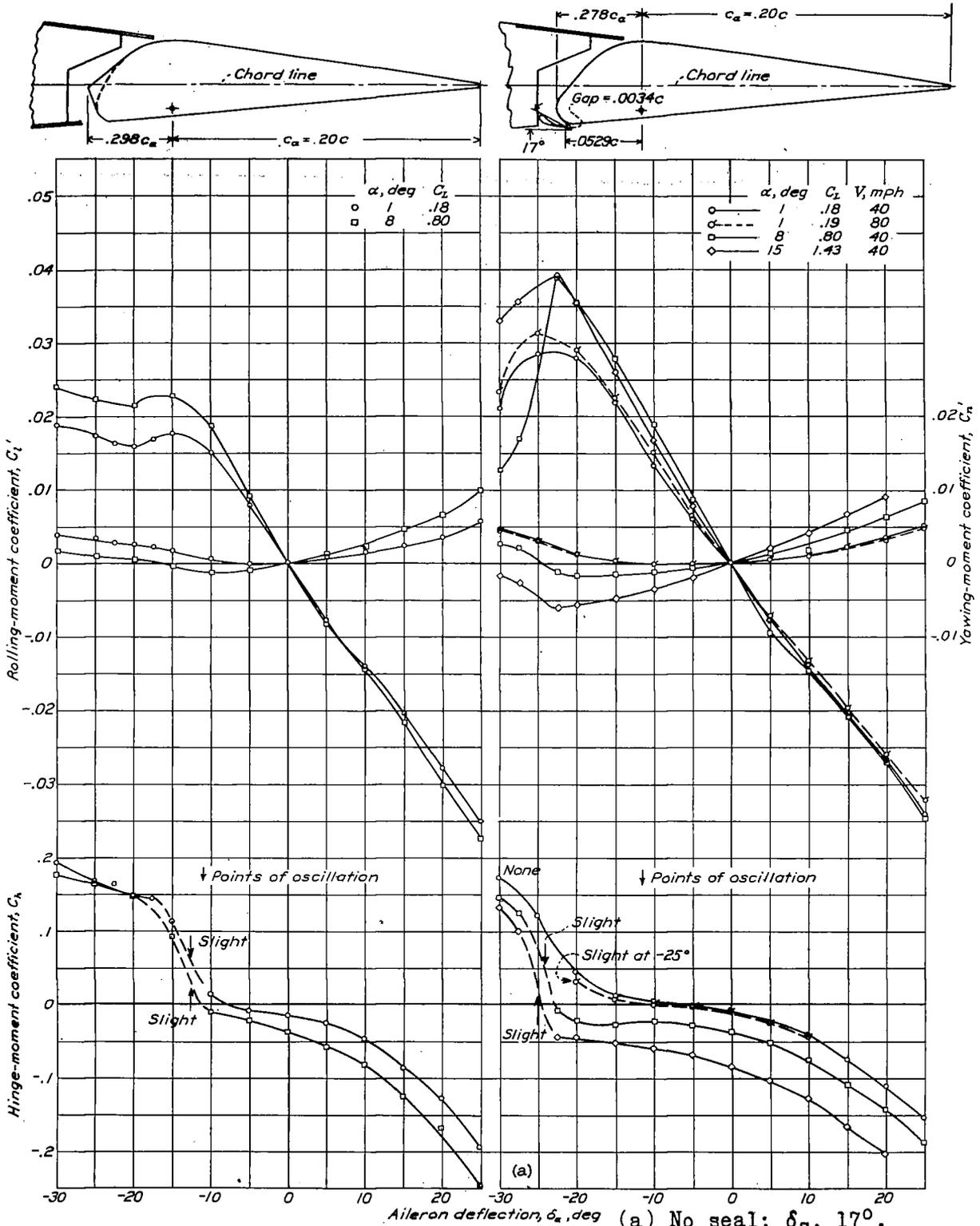
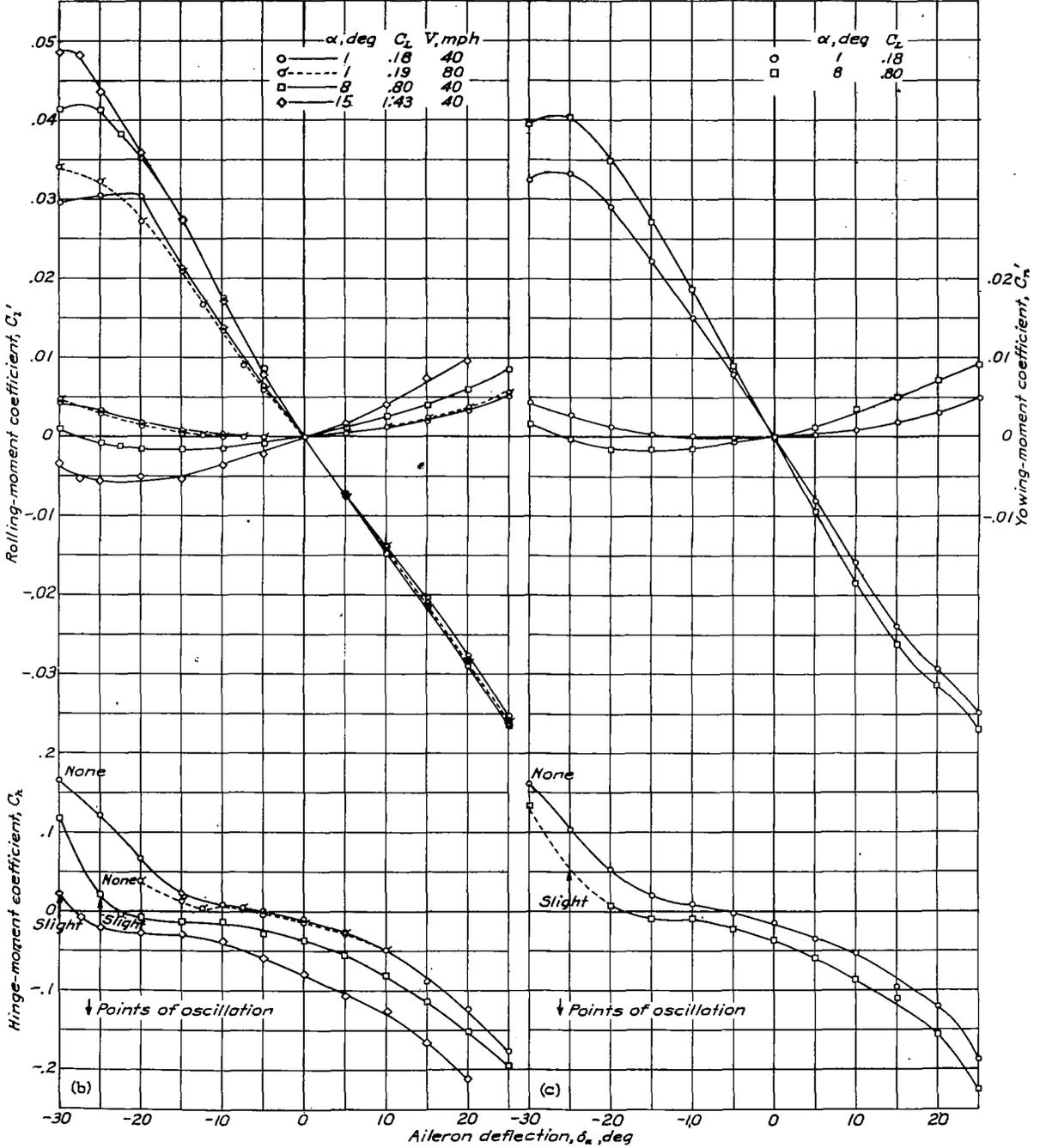
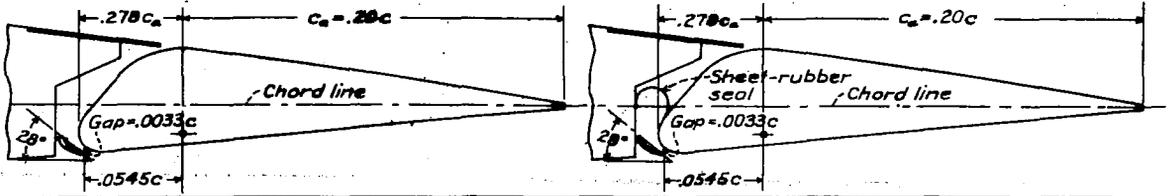


Figure 9.- Aerodynamic characteristics of a 0.20c by 0.37 b/2 Frise aileron with 0.298c<sub>a</sub> balance on an NACA 23012 airfoil. No seal; V, 40 mph.

Figure 10a to c.- Aerodynamic characteristics of a 0.20c by 0.37 b/2 Frise aileron with 0.278c<sub>a</sub> balance and a centrally located 0.078c<sub>a</sub> by 0.31 b/2 nose slat on an NACA 23012 airfoil.



(b) No seal;  $\delta_s, 28^\circ$ .

(c) Sheet-rubber seal;  $\delta_a, 28^\circ$ ; V, 40 mph.

Figure 10b,c.

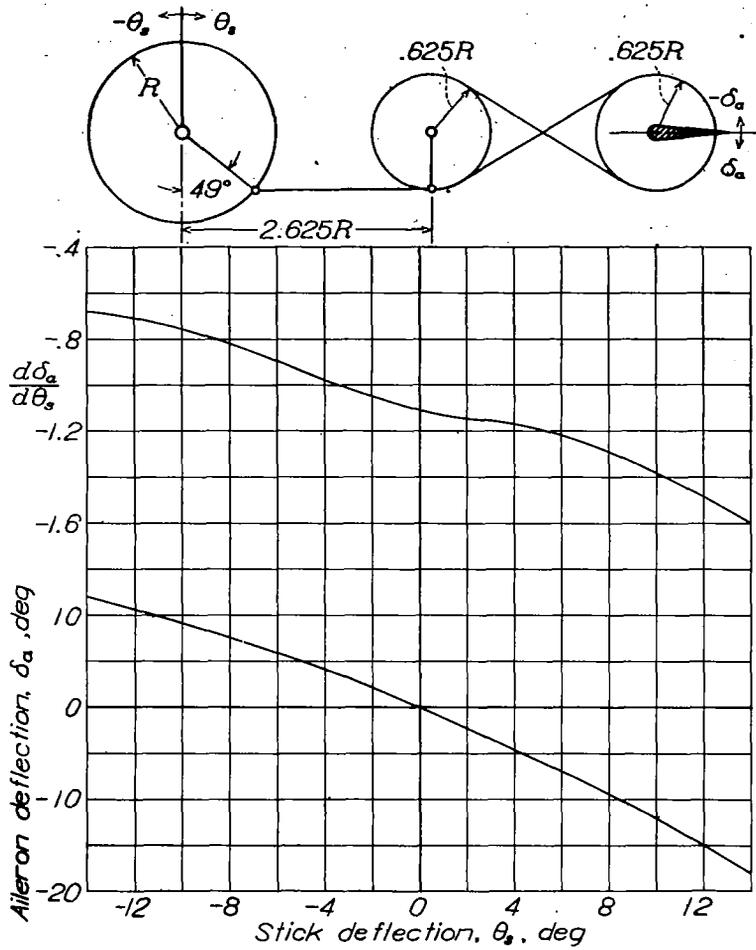
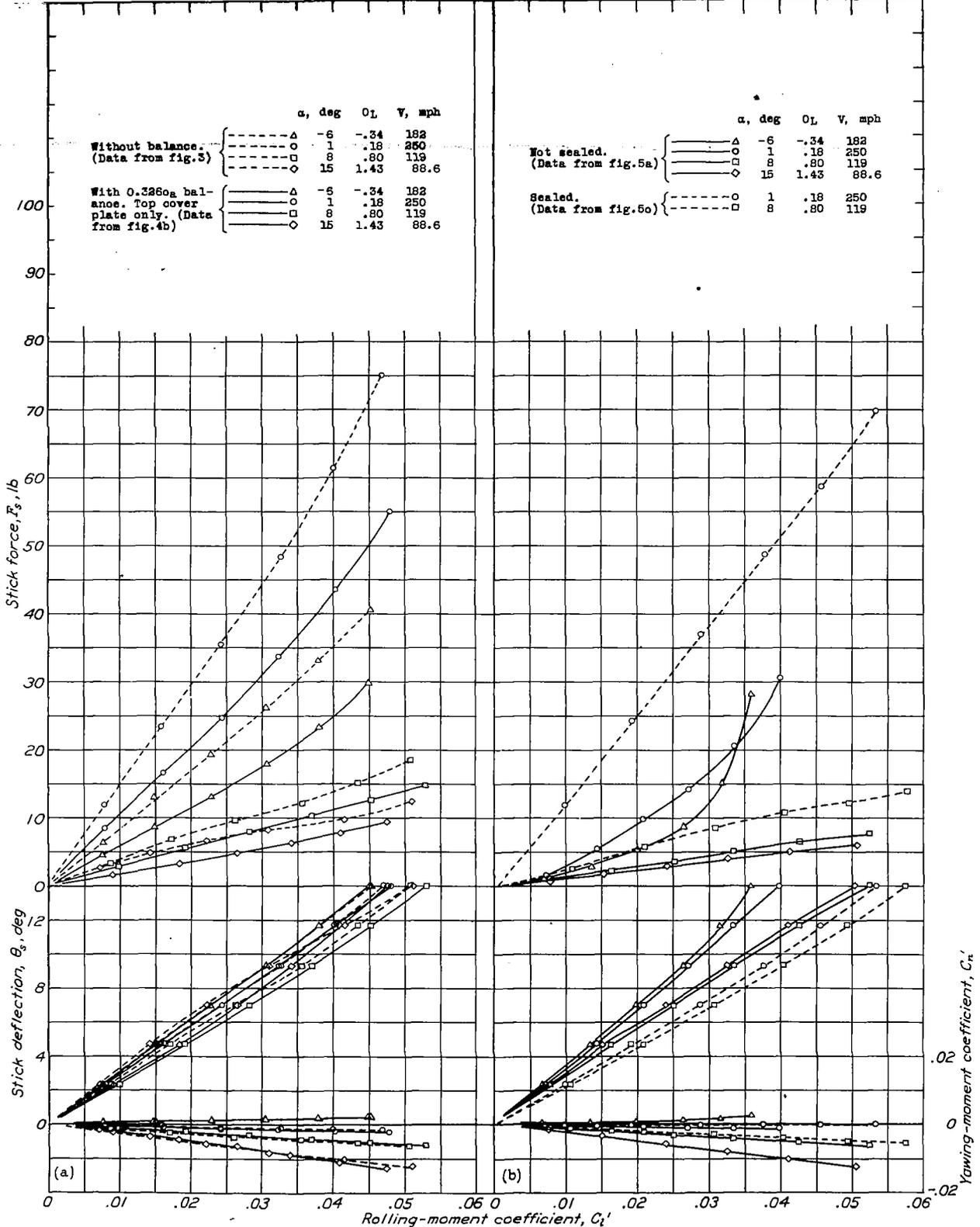
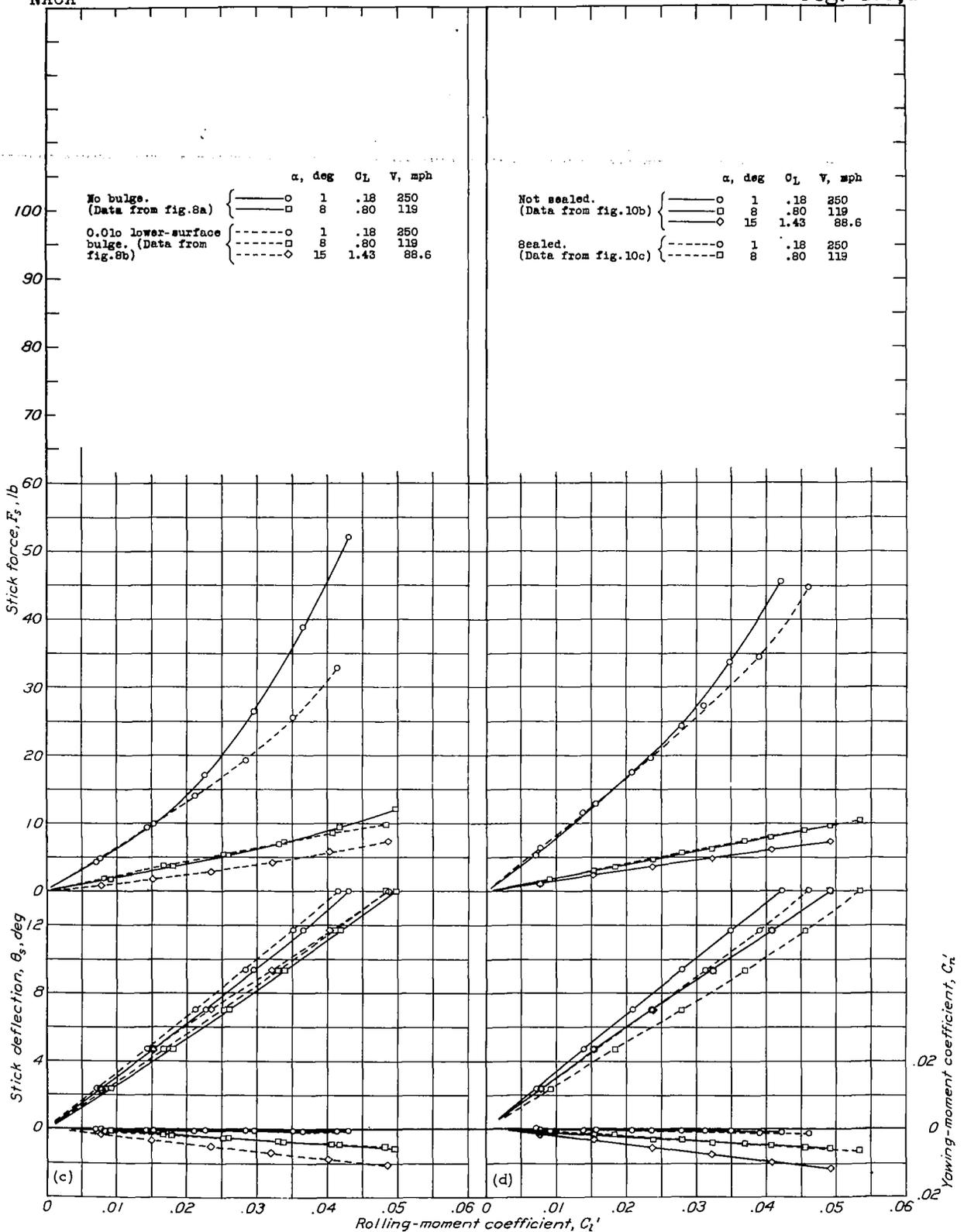


Figure 12.- Conventional differential aileron linkage assumed in the computations.



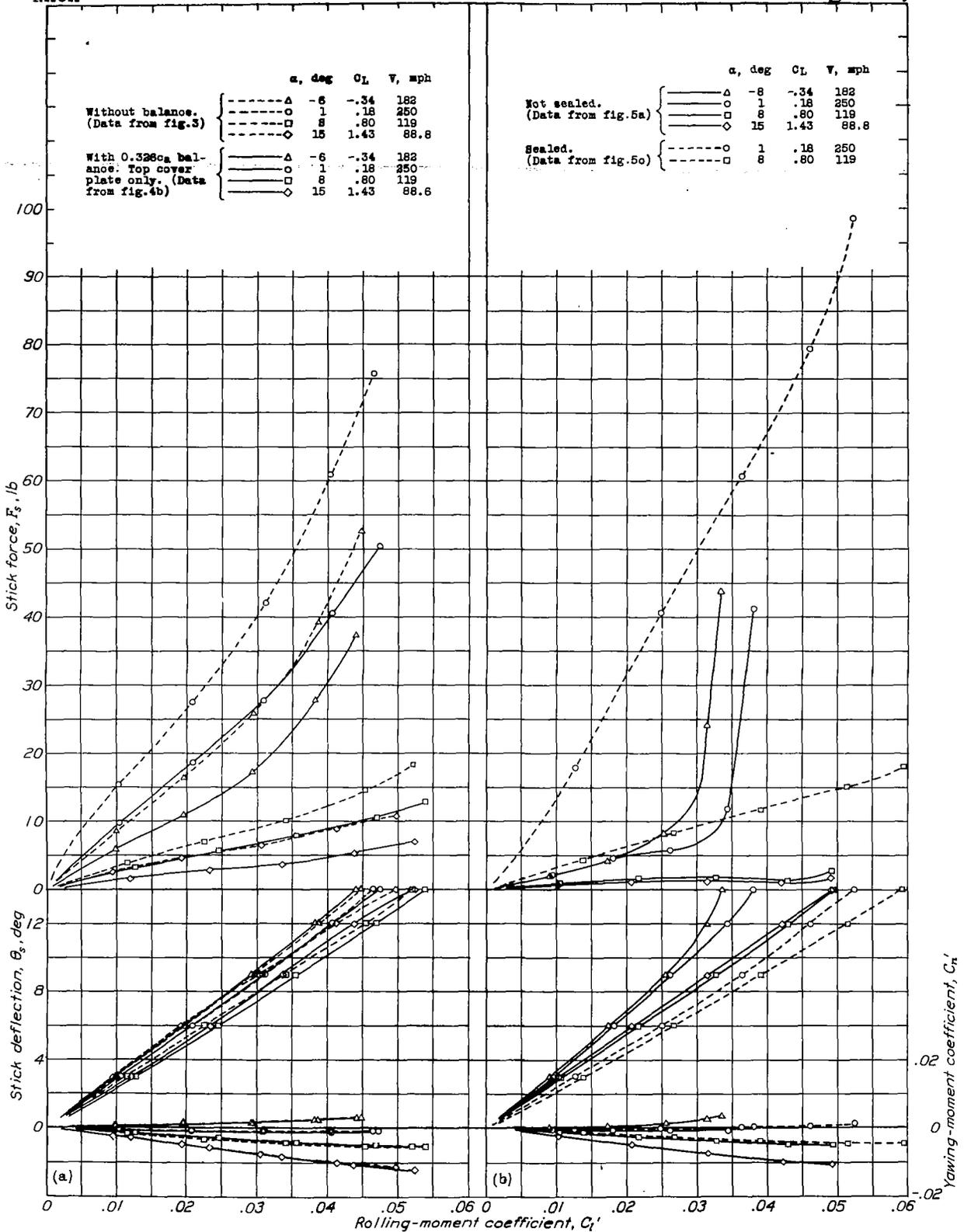
(a) Plain sealed ailerons. (b) Frise aileron with 0.326<sub>a</sub> balance, no bulge.  
 Figure 13a to d.- Aileron-control characteristics of a pursuit airplane equipped with several arrangements of 0.20c by 0.37 b/2 ailerons. Equal up and down linkage.  $\delta_a, \pm 15^\circ$ .



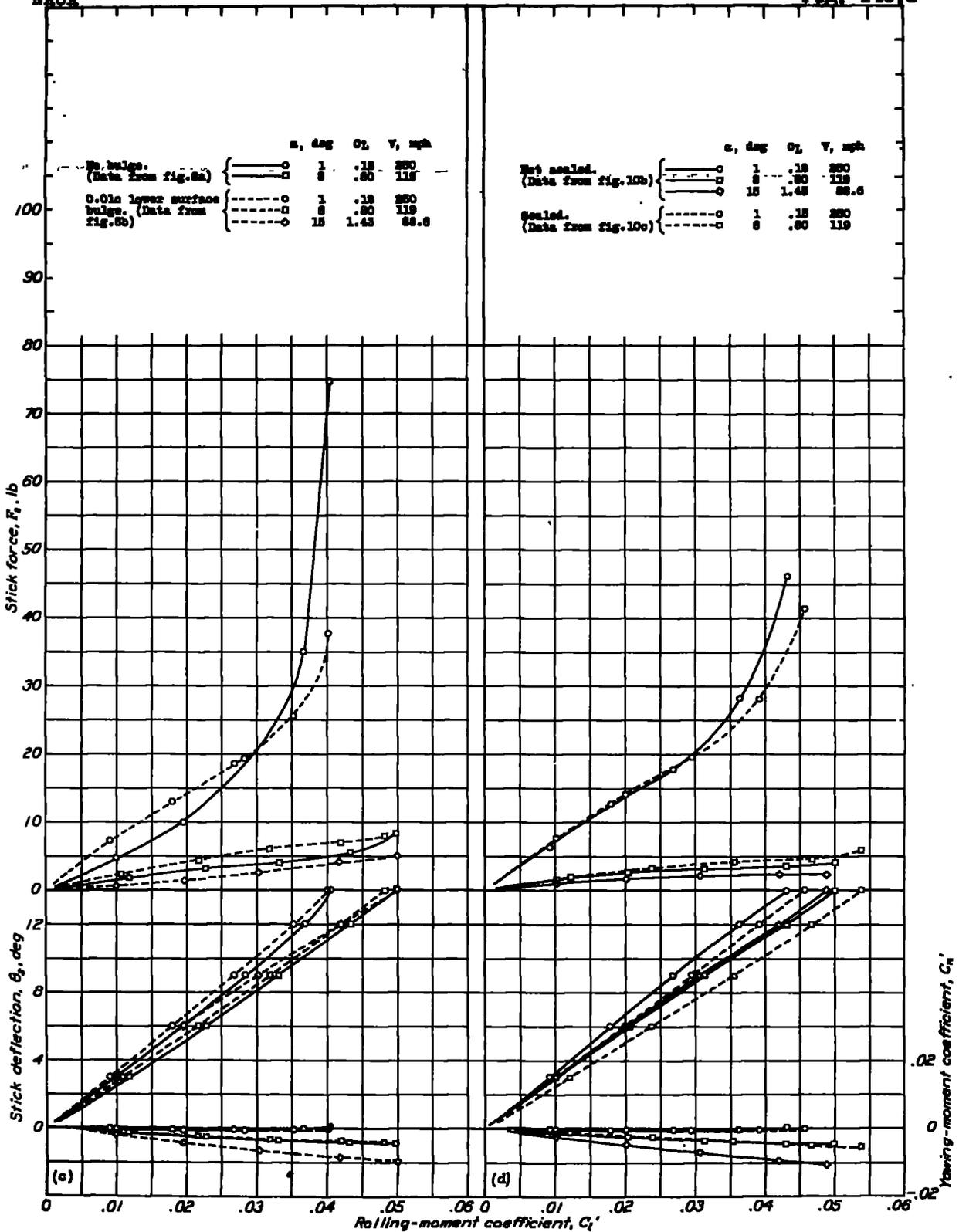
(c) Frise ailerons with 0.278c<sub>a</sub> balance; not sealed.

(d) Frise aileron with 0.278c<sub>a</sub> balance and a nose slat; δ<sub>s</sub>, 28°.

Figure 13c,d.



(a) Plain sealed aileron. (b) Frise aileron with 0.326c<sub>a</sub> balance; no bulge.  
 Figure 14a to d.- Aileron-control characteristics of a pursuit airplane equipped  
 with several arrangements of 0.20c by 0.37 b/2 ailerons.  
 Conventional differential linkage.  $\delta_a, +12^\circ, -18^\circ$ .



(c) Frise ailerons with 0.278c<sub>a</sub> balance; not sealed. (d) Frise aileron with 0.278c<sub>a</sub> balance and a nose slat; δ<sub>B</sub>, 28°

Figure 14c,d.

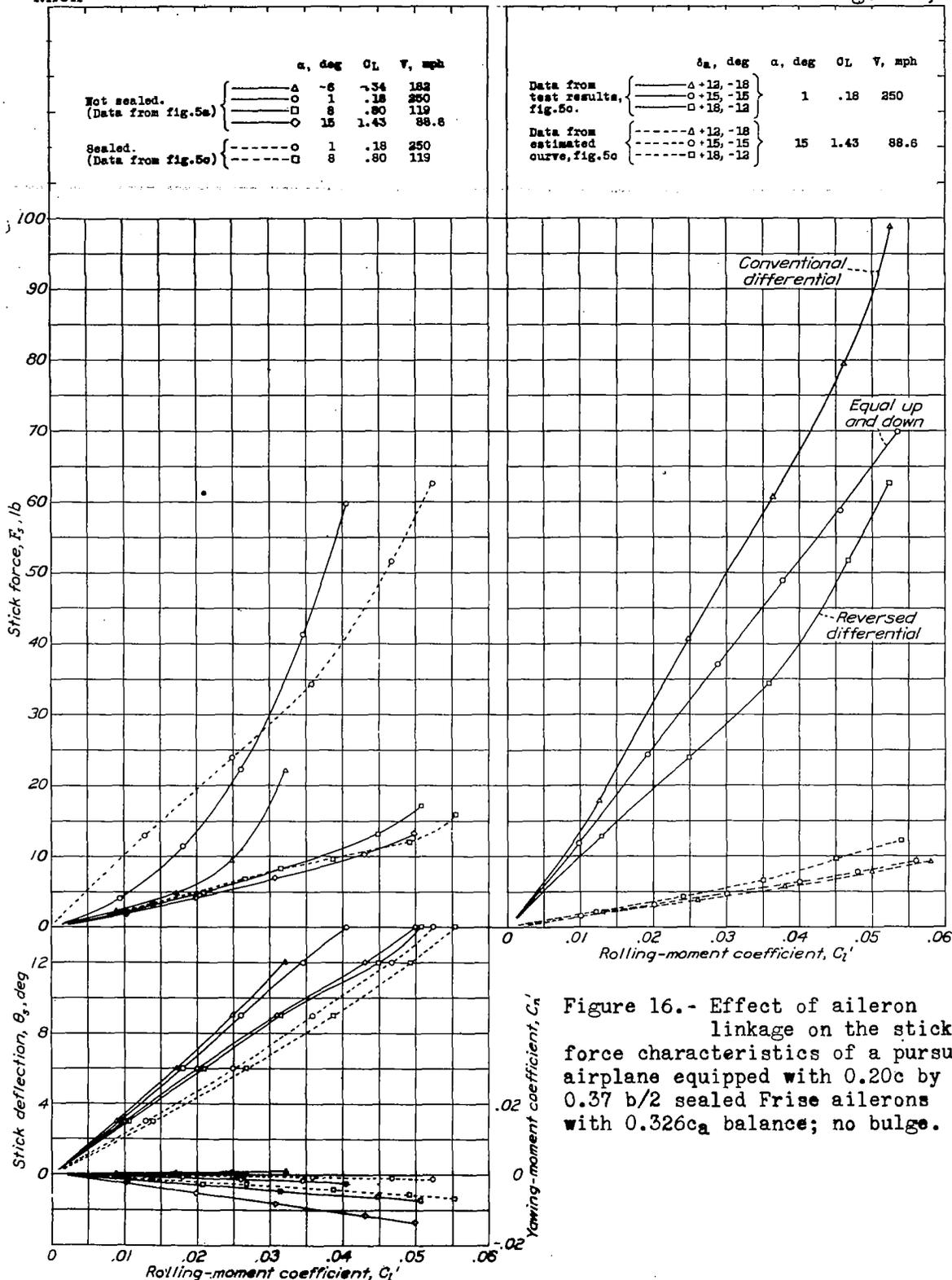
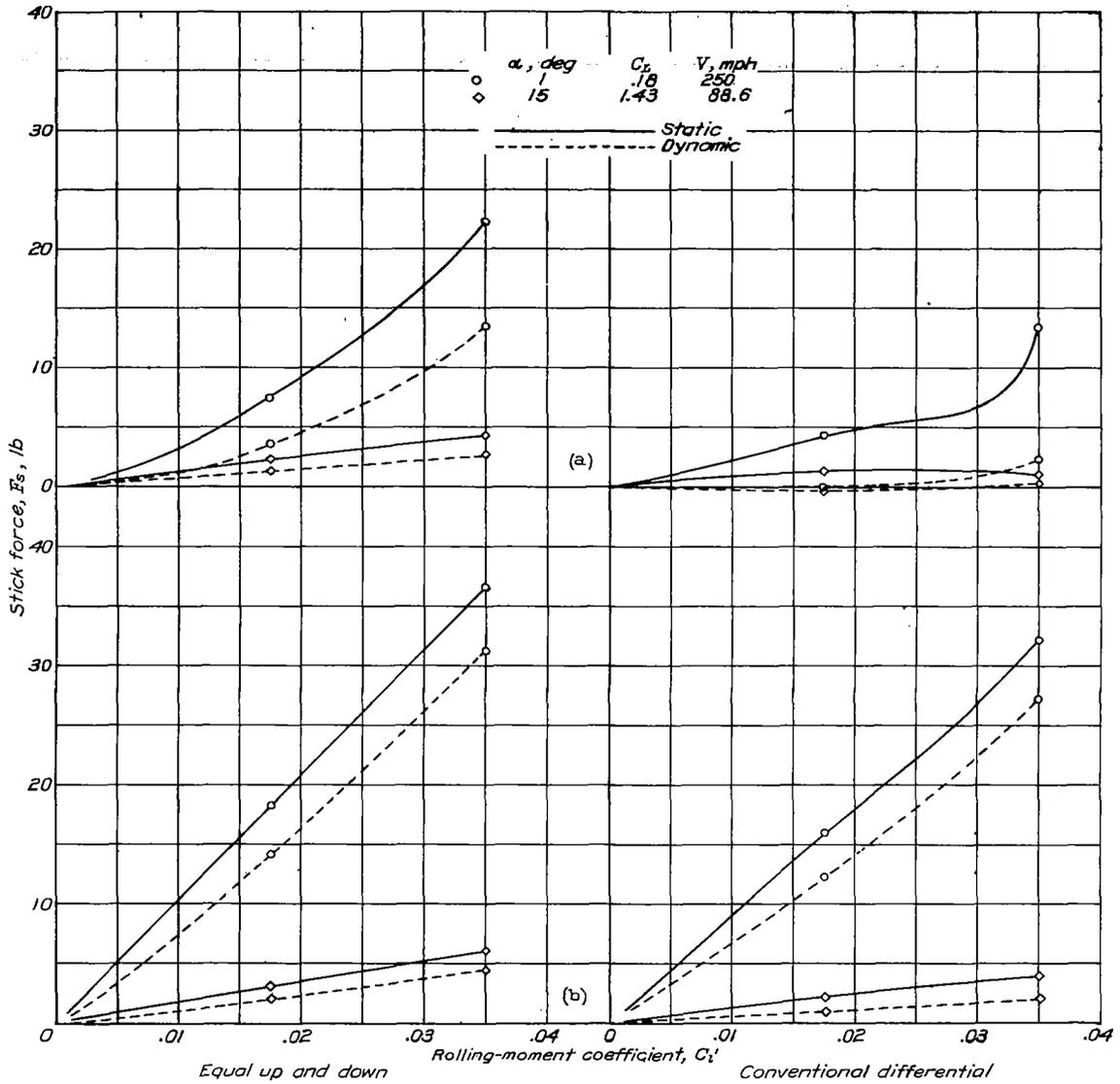


Figure 15.- Aileron-control characteristics of a pursuit airplane equipped with 0.20c by 0.37 b/2 Frise ailerons with 0.326c<sub>a</sub> balance. Reversed differential linkage.  $\delta_a$ , 18°, -12°.

Figure 16.- Effect of aileron linkage on the stick-force characteristics of a pursuit airplane equipped with 0.20c by 0.37 b/2 sealed Frise ailerons with 0.326c<sub>a</sub> balance; no bulge.



(a) Frise aileron with 0.326  $c_m$  balance; no seal, no bulge.

(b) Plain sealed aileron with 0.326  $c_m$  balance; top cover plate only.

Figures 17.- Effect of rolling on the stick-force characteristics of a pursuit airplane equipped with two arrangements of 0.20c by 0.37 b/8 ailerons and two aileron deflection systems.

TITLE: Wind-Tunnel Investigation of 20-Percent-Chord Plain and Frise Ailerons on an  
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ABSTRACT:

Static rolling, yawing, and hinge moments were determined for several angles of attack on an airfoil. Conditions under which the aileron oscillation occurred were also determined. Oscillation of the Frise aileron was the result of abrupt flow breakaway at the lower surface of the aileron nose. Aileron-control characteristics were computed and results are presented illustrating the effects of variation of aileron floating tendency and differential linkage. Internally balanced sealed ailerons with larger amounts of balance than the ailerons which were tested are considered promising.

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