

ARMY RESEARCH LABORATORY



Analytic Model Development for Ceramic Gun Tubes

by Robert Carter

ARL-TR-3648

September 2005

NOTICES

Disclaimers

The findings in this report are not to be construed as an official Department of the Army position unless so designated by other authorized documents.

Citation of manufacturer's or trade names does not constitute an official endorsement or approval of the use thereof.

Destroy this report when it is no longer needed. Do not return it to the originator.

Army Research Laboratory

Aberdeen Proving Ground, MD 21005-5069

ARL-TR-3648

September 2005

Analytic Model Development for Ceramic Gun Tubes

Robert Carter

Weapons and Materials Research Directorate, ARL

REPORT DOCUMENTATION PAGE			<i>Form Approved</i> OMB No. 0704-0188		
Public reporting burden for this collection of information is estimated to average 1 hour per response, including the time for reviewing instructions, searching existing data sources, gathering and maintaining the data needed, and completing and reviewing the collection information. Send comments regarding this burden estimate or any other aspect of this collection of information, including suggestions for reducing the burden, to Department of Defense, Washington Headquarters Services, Directorate for Information Operations and Reports (0704-0188), 1215 Jefferson Davis Highway, Suite 1204, Arlington, VA 22202-4302. Respondents should be aware that notwithstanding any other provision of law, no person shall be subject to any penalty for failing to comply with a collection of information if it does not display a currently valid OMB control number. PLEASE DO NOT RETURN YOUR FORM TO THE ABOVE ADDRESS.					
1. REPORT DATE (DD-MM-YYYY) September 2005		2. REPORT TYPE Final		3. DATES COVERED (From - To) November 2002–November 2004	
4. TITLE AND SUBTITLE Analytic Model Development for Ceramic Gun Tubes			5a. CONTRACT NUMBER		
			5b. GRANT NUMBER		
			5c. PROGRAM ELEMENT NUMBER		
6. AUTHOR(S) Robert Carter			5d. PROJECT NUMBER 622618H80		
			5e. TASK NUMBER		
			5f. WORK UNIT NUMBER		
7. PERFORMING ORGANIZATION NAME(S) AND ADDRESS(ES) U.S. Army Research Laboratory ATTN: AMSRD-ARL-WM-MB Aberdeen Proving Ground, MD 21005-5069			8. PERFORMING ORGANIZATION REPORT NUMBER ARL-TR-3648		
9. SPONSORING/MONITORING AGENCY NAME(S) AND ADDRESS(ES)			10. SPONSOR/MONITOR'S ACRONYM(S)		
			11. SPONSOR/MONITOR'S REPORT NUMBER(S)		
12. DISTRIBUTION/AVAILABILITY STATEMENT Approved for public release; distribution is unlimited.					
13. SUPPLEMENTARY NOTES					
14. ABSTRACT The equations for calculating the probability of failure for an internally pressurized ceramic tube are straightforward and well defined in the literature. The most common solution is for a tube subjected only to an internal pressure. This approach neglects any external pressure, over wrap, or sheathing system that may be used to maintain or promote a beneficial compressive pre-stress. The current research has focused on the development of a model, based on elasticity theory, of a sheathed ceramic tube augmented with probability of failure calculations used to calculate stress, strain, and probability of failure of the ceramic tube component. The model calculated the Weibull probability of failure for volumetric- and surface-strength-limiting flaws with the consideration of predicted (isothermal, mechanical, or combined) stress states on the surfaces and throughout the volume.					
15. SUBJECT TERMS ceramic gun tubes, Weibull statistics, failure surface					
16. SECURITY CLASSIFICATION OF:			17. LIMITATION OF ABSTRACT UL	18. NUMBER OF PAGES 30	19a. NAME OF RESPONSIBLE PERSON Robert Carter
a. REPORT UNCLASSIFIED	b. ABSTRACT UNCLASSIFIED	c. THIS PAGE UNCLASSIFIED			19b. TELEPHONE NUMBER (Include area code) 410-306-1102

Contents

Acknowledgments	iv
1. Introduction	1
1.1 Statistical Model Development	1
1.2 Volume Flows.....	2
1.3 Surface Flaws	3
1.4 Total Probability of Failure	4
2. Sheathed Tubes	4
2.1 Sheathed Tube Mechanics.....	5
2.2 Failure Surfaces for Pressurized Sheathed Tubes	8
3. Summary	10
4. References	11
Distribution List	12

Acknowledgments

The author would like to acknowledge Dr. Andy Wereszczak for his help and guidance with this work. Also, this research was supported in part by an appointment to the Research Participation Program at the U.S. Army Research Laboratory (ARL) administered by the Oak Ridge Institute for Science and Education through an interagency agreement between the U.S. Department of Energy and ARL.

1. Introduction

Ceramic materials are being investigated as bore materials for high performance gun barrels. The superior high temperature behavior and excellent erosion resistance make them strong candidates for application in the harsh environment produced during a ballistic event. The application of these materials is not a novel approach. Several previous investigations by the U.S. Army, Navy, and different contract organizations have considered the use of ceramic materials in gun bores. The success of these previous investigations has been limited at best, but advances in ceramic material processing, probabilistic design, and sheathing technologies have prompted the U. S. Army Research Laboratory to renew investigations into ceramic materials.

The primary factor preventing the simple insertion of ceramics into gun bores has been the inability to design around the low tensile strength, large variability in the observed strength, and brittle behavior of the materials. The objective of the present research is to develop analytic models for the design of ceramic-lined gun barrels capable of surviving interior ballistic events. The first section of this report will focus on the derivation of the statistical equations for predicting failure in ceramics, while the second will deal with the development of a model for calculating stress. The results of the completed model will be presented showing the usefulness of the model.

1.1 Statistical Model Development

Statistical methods are necessary to properly design around the variability in observed strength of ceramic components. The most recognized approach incorporates the Weibull distribution equation. The original Weibull equation calculates a probability of failure (P_f) for a brittle material subjected to a uniaxial stress distribution:

$$P_f = 1 - e^{-\int \left(\frac{\sigma}{\sigma_o}\right)^m dV}, \quad (1)$$

with σ being the stress, σ_o is the Weibull strength or scale parameter, and m being the Weibull modulus. This expression considers only one type of flaw population located in the volume of the ceramic body subjected to a uniform stress (I). For a pressurized tube, additional conditions need to be evaluated, namely the probability for a nonuniform stress state and multiple flaw populations. This report will derive the equations needed to calculate the probability of failure due to volume and surface flaws for a tube subjected to internal and external pressures only. Failure calculations evaluate the nonuniform hoop tensile stresses only and assume that the probability of failure of a part in compression is zero (as described in ASTM Standard C 1239-00) (2). The calculations also assume that the tube is loaded along its entire length, thus eliminating the consideration of any edge effects.

1.2 Volume Flaws

For the condition when the stress distribution is unidirectional, but not uniform,

$$P_{fV} = 1 - e^{-k_v V \left(\frac{\sigma_{\max}}{\sigma_{oV}} \right)^{m_v}}, \quad (2)$$

where

$$k_v V = \int \left(\frac{\sigma(r)}{\sigma_{\max}} \right)^{m_v} dV, \quad (3)$$

σ_{\max} is the maximum stress in the material, $\sigma(r)$ is the function describing the stress distribution, m_v is the volumetric Weibull modulus, and $k_v V$ is the effective volume of the sample (3). Some distinction needs to be placed on the Weibull material scale parameter (σ_o) since it is not always the same as the Weibull characteristic strength (often listed as σ_θ — but not to be confused with hoop stress in this report). The characteristic strength term, σ_θ , is often reported, but is test and sample geometry dependant. The volumetric material scale parameter, σ_{oV} , is strength per unit volume of uniform, uniaxial tension (hence the unusual units of MPa*mm^{3/m_v}).

For a tube subjected to internal and external pressures, the Lamé cylinder expression for the hoop stress (σ_θ) is

$$\sigma_\theta(r) = \frac{P_i r_i^2 - P_o r_o^2}{r_o^2 - r_i^2} - \frac{r_i^2 r_o^2 (P_o - P_i)}{r_o^2 - r_i^2} \frac{1}{r^2}, \quad (4)$$

where r is radius, P is the pressure, and i and o refer to the inner and outer surfaces, respectively (4). This distribution exhibits maximum stress at the inner surface when $P_i > P_o$, and is given by

$$\sigma_{\theta\max} = \frac{P_i (r_i^2 + r_o^2) - 2P_o r_o^2}{r_o^2 - r_i^2}. \quad (5)$$

The effective volume, $k_v V$, for a tube subjected to internal and external pressures is found by substituting equations 4 and 5 into 3:

$$k_v V = 2\pi L \int_{r_i}^{r_o} \left[\frac{P_i r_i^2 - P_o r_o^2}{P_i (r_i^2 + r_o^2) - 2P_o r_o^2} + \frac{r_i^2 r_o^2 (P_i - P_o)}{P_i (r_i^2 + r_o^2) - 2P_o r_o^2} \frac{1}{r^2} \right]^{m_v} r dr. \quad (6)$$

It should be noted that including the external pressure condition creates three regimes of behavior: P_i dominant, transition, and P_o dominant. In the P_i dominant condition, where $P_i \gg P_o$, the hoop stress is tensile through the thickness, and exhibits a maximum at the inner

surface. In the P_o dominant case, where $P_o \gg P_i$, it is compressive throughout the thickness. In the transition regime, where $P_i > P_o$, the inner portion of the tube is tension, while the outer is in compression.

The limits for the different dominant behaviors are determined by setting the hoop stress terms to zero at both surfaces and solving for the external pressure:

$$\begin{aligned} P_{otrans} &= \frac{2P_i r_i^2}{r_i^2 + r_o^2} , \\ P_{ocompression} &= \frac{P_i(r_i^2 + r_o^2)}{2r_o^2} . \end{aligned} \quad (7)$$

P_{otrans} is the solution for when the outer surface of the tube goes into compression, while $P_{ocompression}$ is when the entire tube is in compression. In the transition region, the radial position where the hoop stress is zero is necessary for evaluating $k_V V$. This location, termed r_{neu} , is given by

$$r_{neu} = \frac{-r_i^2 r_o^2}{(P_i r_i^2 - P_o r_o^2)} \sqrt{(P_i r_i^2 - P_o r_o^2)(P_o - P_i)} . \quad (8)$$

When $P_{otrans} < P_o < P_{ocompression}$, a portion of the tube is in compression (r_i to r_{neu}), so $k_V V$ changes to

$$k_V V = 2\pi L \int_{r_i}^{r_{neu}} \left[\frac{P_i r_i^2 - P_o r_o^2}{P_i(r_i^2 + r_o^2) - 2P_o r_o^2} + \frac{r_i^2 r_o^2 (P_i - P_o)}{P_i(r_i^2 + r_o^2) - 2P_o r_o^2} \frac{1}{r^2} \right]^{m_v} r dr . \quad (9)$$

As stated earlier, if $P_o > P_{ocompression}$, then $k_V V = 0$.

With the $k_V V$ term, the expression in equation 2 can be solved for the P_f value for volume flaws.

1.3 Surface Flaws

The expression for the probability of failure is similar to that of the volume flaws:

$$P_{fA} = 1 - e^{-k_{fA} A \left(\frac{\sigma_{\theta \max}}{\sigma_{oA}} \right)^{m_A}} , \quad (10)$$

however, due to the population of flaws located at surface the integral for the effective area, k_{fA} operates over the surface area, not the volume. The effective area is evaluated over the inner surface, the two ends, and the outer surface as shown in equation 11:

$$\begin{aligned}
k_{AA} = & 2\pi L r_i + 4\pi \int_{r_i}^{r_o} \left[\frac{P_i r_i^2 - P_o r_o^2}{P_i (r_i^2 + r_o^2) - 2P_o r_o^2} + \frac{r_i^2 r_o^2 (P_i - P_o)}{P_i (r_i^2 + r_o^2) - 2P_o r_o^2} \frac{1}{r^2} \right]^{m_A} r dr \\
& + 2\pi L r_o \left[\frac{2P_i r_i^2 - P_o (r_i^2 + r_o^2)}{(r_i^2 + r_o^2) P_i - 2P_o r_o^2} \right]^{m_A}.
\end{aligned} \tag{11}$$

In the case when $P_{otrans} < P_o < P_{ocompression}$, the k_{AA} expression changes to

$$k_{AA} = 2\pi L r_i + 4\pi \int_{r_i}^{r_{neu}} \left[\frac{P_i r_i^2 - P_o r_o^2}{P_i (r_i^2 + r_o^2) - 2P_o r_o^2} + \frac{r_i^2 r_o^2 (P_i - P_o)}{P_i (r_i^2 + r_o^2) - 2P_o r_o^2} \frac{1}{r^2} \right]^{m_A} r dr \quad . \tag{12}$$

Since the outer surface is in compression the outer surface term is dropped and the integral is evaluated on the tensile region. If $P_o > P_{ocompression}$, then $k_{AA} = 0$.

1.4 Total Probability of Failure

The two expressions for probability of failure, one for volume flaws and one for surface flaws, are combined to calculate the probability of failure for the tube (5):

$$P_f = 1 - \prod_{i=1}^N (1 - P_{fi}) \quad , \tag{13}$$

where P_f is the total probability of failure for the component and P_{fi} is the probability of failure for the i^{th} flaw population or location. Given the two Weibull parameters (for each flaw population), the new P_f value can be calculated for a combination of internal and external pressure. Also, more terms can be added to equation 13 to address multiple flaw populations or to address each flaw type encountered.

2. Sheathed Tubes

In order to develop a gun system capable of withstanding the pressure loads induced by the ballistic firing, sheathing material needs to be applied in a way to generate beneficial compressive pre-stress. The level of the pre-stress is highly dependant upon the elastic properties of the sheath and ceramic, the strengths of both materials, and the thermal expansion coefficients. Analytic models are available to describe the response of tubes to the loading conditions that can be expected from a gun system.

2.1 Sheathed Tube Mechanics

The simplest model for sheathed tubes is that of two isotropic materials, which is adequate for a metal-sheathed ceramic tube. The equations located in the text by Herakovich and the works by Rousseau and Hyer provide good guidance to calculating the stress, strain, and displacement relations for a system of axisymmetric, nested tubes subjected to uniform internal and external pressure, axial tension and compression, axial torsion, and uniform temperature changes (6–8). They are sufficient to model interference stresses from shrink-fit and press-fit operations which are needed for imparting a beneficial pre-stress into the ceramic.

The most well known and simplest expression for stress in an isotropic tube due to internal and external pressure is the Lamé cylinder expression:

$$\begin{aligned}\sigma_{\theta}(r) &= -\frac{a^2 b^2 (p_o - p_i)}{(b^2 - a^2)} \frac{1}{r^2} + \frac{p_i a^2 - p_o b^2}{(b^2 - a^2)} \\ \sigma_r(r) &= \frac{a^2 b^2 (p_o - p_i)}{(b^2 - a^2)} \frac{1}{r^2} + \frac{p_i a^2 - p_o b^2}{(b^2 - a^2)},\end{aligned}\quad (14)$$

where

$$\begin{aligned}\sigma_{\theta} &= \text{hoop stress} \\ a &= \text{inner radius} \\ b &= \text{outer radius} \\ p_o &= \text{external pressure} \\ p_i &= \text{internal pressure} \\ r &= \text{radial location.}\end{aligned}$$

This expression is useful for calculating the stresses in a monolithic ring with an external pressure representing the effects of the sheath. Simulations using this approach are useful for determining the effects of varying the material properties, but it is not effective for simulating a sheathed system since the external pressure on the ceramic is constant.

In order to accurately model a sheathed system, a more comprehensive model is necessary. In the chapter on laminated tubes in the text by Herakovich, expressions for a layered tube are developed. The following expressions use the cylindrical coordinate system labeled x (axial), θ (circumferential), and r (radial) directions.

For an isotropic material, the axial, tangential, and radial displacements, $u(x)$, $v(x, r)$, and $w(r)$ respectively, are defined as

$$\begin{aligned}u(x) &= \varepsilon_x^o x \\ v(x, r) &= \gamma^o x r \\ w(r) &= A_1 r + A_2 r^{-1},\end{aligned}\quad (15)$$

where $A_1, A_2, \varepsilon_x^o$, and γ^o are unknown constants. Similarly, the expressions for anisotropic materials are identical for $u(x)$ and $v(x,r)$, but $w(r)$ becomes

$$\begin{aligned}
 w(r) &= A_1 r^\lambda + A_2 r^{-\lambda} + \Gamma \varepsilon_x^o r + \Omega \gamma^o r^2 + \Psi r \Delta T \\
 \Gamma &= \left(\frac{\bar{C}_{12} - \bar{C}_{13}}{\bar{C}_{33} - \bar{C}_{22}} \right) \\
 \Omega &= \left(\frac{\bar{C}_{26} - 2\bar{C}_{36}}{4\bar{C}_{33} - \bar{C}_{22}} \right) \\
 \Psi &= \left(\frac{\tilde{\Sigma}}{\bar{C}_{33} - \bar{C}_{22}} \right) \\
 \tilde{\Sigma} &= \sum_i (\bar{C}_{i3} - \bar{C}_{i2}) \alpha_i, \tag{16}
 \end{aligned}$$

where \bar{C}_{ij} is a transformed stiffness matrix value.

For a single ply, there are the four unknowns for which to solve, but the number of unknown scales for laminates of more than one ply. There are single values for ε_x^o and γ^o , but there are values for A_1 and A_2 for every layer. For a structure with N layers, this translates to a total number of unknowns of $2N + 2$.

The first step to solve for the unknown values is to transform the equations for displacement to the strain and stress relations. This allows for the use of the stress and strain boundary conditions to help define the unknown values. First consider the strain-displacement relations for cylindrical coordinates:

$$\begin{aligned}
 \varepsilon_x &= \frac{\partial u}{\partial x} & \gamma_{r\theta} &= \frac{\partial v}{\partial r} - \frac{v}{r} \\
 \varepsilon_\theta &= \frac{w}{r} & \gamma_{xr} &= \frac{\partial u}{\partial r} \\
 \varepsilon_r &= \frac{\partial w}{\partial r} & \gamma_{x\theta} &= \frac{\partial v}{\partial x}. \tag{17}
 \end{aligned}$$

Substituting equations 15–17, the strains can be written in terms of the unknown values. Using the three-dimensional constitutive equations in cylindrical coordinates, the stress expression can be derived from the strain equations:

$$\begin{bmatrix} \sigma_x \\ \sigma_\theta \\ \sigma_r \\ \tau_{\theta r} \\ \tau_{xr} \\ \tau_{x\theta} \end{bmatrix} = \begin{bmatrix} \bar{C}_{11} & \bar{C}_{12} & \bar{C}_{13} & 0 & 0 & \bar{C}_{16} \\ \bar{C}_{12} & \bar{C}_{22} & \bar{C}_{23} & 0 & 0 & \bar{C}_{26} \\ \bar{C}_{13} & \bar{C}_{23} & \bar{C}_{33} & 0 & 0 & \bar{C}_{36} \\ 0 & 0 & 0 & \bar{C}_{44} & \bar{C}_{45} & 0 \\ 0 & 0 & 0 & \bar{C}_{45} & \bar{C}_{55} & 0 \\ \bar{C}_{16} & \bar{C}_{26} & \bar{C}_{36} & 0 & 0 & \bar{C}_{66} \end{bmatrix} \begin{bmatrix} \varepsilon_x \\ \varepsilon_\theta \\ \varepsilon_r \\ \gamma_{\theta r} \\ \gamma_{xr} \\ \gamma_{x\theta} \end{bmatrix}, \tag{18}$$

where σ and τ are normal and shear stresses, \bar{C}_{ij} is the transformed stiffness matrix, and ε and γ are the normal and shear strains. Thermal and filament winding stresses can be included with a modification to the constitutive equations. If the strain values are modified to include thermal and winding strain values, as in

$$\varepsilon = \varepsilon^E - \varepsilon^{Th} + \varepsilon^w \quad , \quad (19)$$

where ε is the total strain, ε^E is the elastic strain, ε^{Th} is the thermal strain ($\alpha\Delta T$), and ε^w is the elastic strain in the tow due to winding tension. Winding strain is the elastic strain imparted due to tow tension in winding, or

$$\varepsilon^w = \frac{F}{AE} \quad , \quad (20)$$

where F is the force on the tow, A is the area of the tow, and E is the elastic modulus of the fibers (9).

With the equations for stress, strain, and displacement, the unknowns can be found by applying different boundary conditions. There are force and torque conditions that must be met:

$$\begin{aligned} F_x &= 2\pi \sum_{k=1}^N \int_{r_{k-1}}^{r_k} \sigma_x^{(k)} r dr \\ T_x &= 2\pi \sum_{k=1}^N \int_{r_{k-1}}^{r_k} \tau_{x\theta}^{(k)} r^2 dr \quad . \end{aligned} \quad (21)$$

The applied axial force (F_x) and axial torque (T_x) are equal to the sum of the axial and shear stresses integrated over the area of each layer. This provides two equations for the $2N + 2$ unknowns.

Two more equations come from the balance of stresses at the inner and outer surfaces of the tube. The radial stress at each surface must balance the applied pressure, or

$$\begin{aligned} -p_i &= \sigma_r^1(R_i) \\ -p_o &= \sigma_r^N(R_o) \quad . \end{aligned} \quad (22)$$

The final $2N - 2$ equations come from continuity of traction and displacement at each internal interface. The radial stresses and displacements must be continuous across each interface, so

$$\begin{aligned} w^{(k)}(r_k) &= w^{(k+1)}(r_k) \\ \sigma_r^{(k)}(r_k) &= \sigma_r^{(k+1)}(r_k) \quad . \end{aligned} \quad (23)$$

There are now $2N + 2$ equations and $2N + 2$ unknowns, so the system can be solved for a given loading conditions.

2.2 Failure Surfaces for Pressurized Sheathed Tubes

The stress relations are used to calculate the stress profile through the wall of the ceramic tube. The values are the input into the probability of failure expressions derived in the previous sections, and the probability of failure for a pressurized tube is calculated. Mathcad* software was used to solve for variations in the material properties of the sheath and ceramic, geometry of the tubes, and operating conditions. With the implementation of a failure criterion for the sheath, the model can be used for designing optimal pre-stress generation with failure of the different materials.

A good example of this problem is the effect of the change in temperature has on the volumetric P_f of a steel-sheathed silicon nitride tube with an internal pressure of 500 MPa. The material properties are listed in table 1, and the failure curves in figure 1 are for a tube with an inner diameter (ID) of 10 cm, and outer diameter (OD) of 20 cm, length of 1 m. The thickness of the ceramic was varied from 2.5, 5, and 7.5 cm with the remaining portion of the 10-cm thickness being steel. Due to the thermal expansion mismatch between steel ($\alpha = 12.8 \text{ ppm}/^\circ\text{C}$) and silicon nitride ($\alpha = 3 \text{ ppm}/^\circ\text{C}$), cooling the tube assembly will generate compressive stresses in the ceramic. The probability of failure decreases as the ΔT value becomes more negative.

Table 1. Material properties for steel and silicon nitride.

Property	Steel	Si ₃ N ₄
Modulus (GPa)	200	310
Poisson's ratio	0.32	0.24
CTE (ppm/°C)	12.8	3
σ_{0V} (MPa*mm ^{3/m})	—	1190
m_v	—	25

The probability of failure is plotted on a logarithmic scale to illustrate the behavior when the values become increasingly small. The thinner ceramic wall has a larger probability of failure at a small ΔT , but surpasses the thicker ceramic assemblies between -100 and -200 °C.

A more informative method of displaying the effects of varying ceramic wall thickness and pre-stress levels is to create failure surfaces for the sheath and ceramic materials (10). This is accomplished by fixing the total wall thickness and varying the ceramic-to-sheath ratio. Also, the pre-stress level can be varied from an unstressed condition to a large magnitude stress. Failure will be calculated by the probability of failure of the ceramic and by a yield failure criterion for the sheath. The resulting plot is illustrated in figure 2. The x -axis is the change in temperature from when the sheath makes initial contact with the ceramic for a shrink-fit operation. The y -axis is the ratio of the ceramic wall thickness to total wall thickness (0% is a steel tube with no ceramic and 100% is all ceramic with no sheath). The color codes are for the log of the probability of failure—zero is a P_f of 100% or zero chance of success, negative six is a

*Mathcad is a registered trademark of Mathsoft.

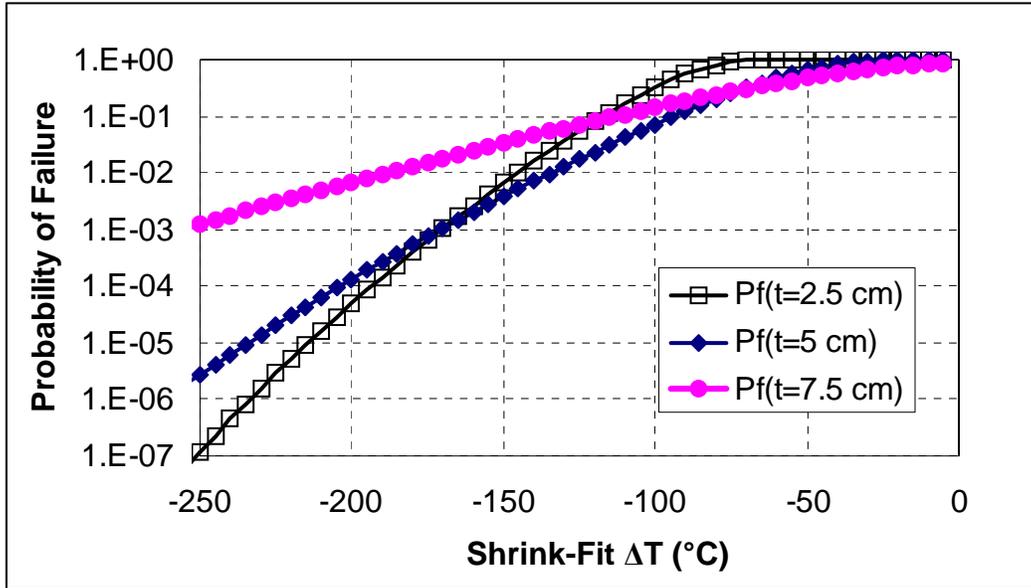


Figure 1. Probability of failure for a steel-sheathed silicon nitride tube with an internal pressure of 500 MPa. The three curves are for different ceramic wall thickness for a tube with an ID of 10 cm, an OD of 20 cm, and a length of 1 m.

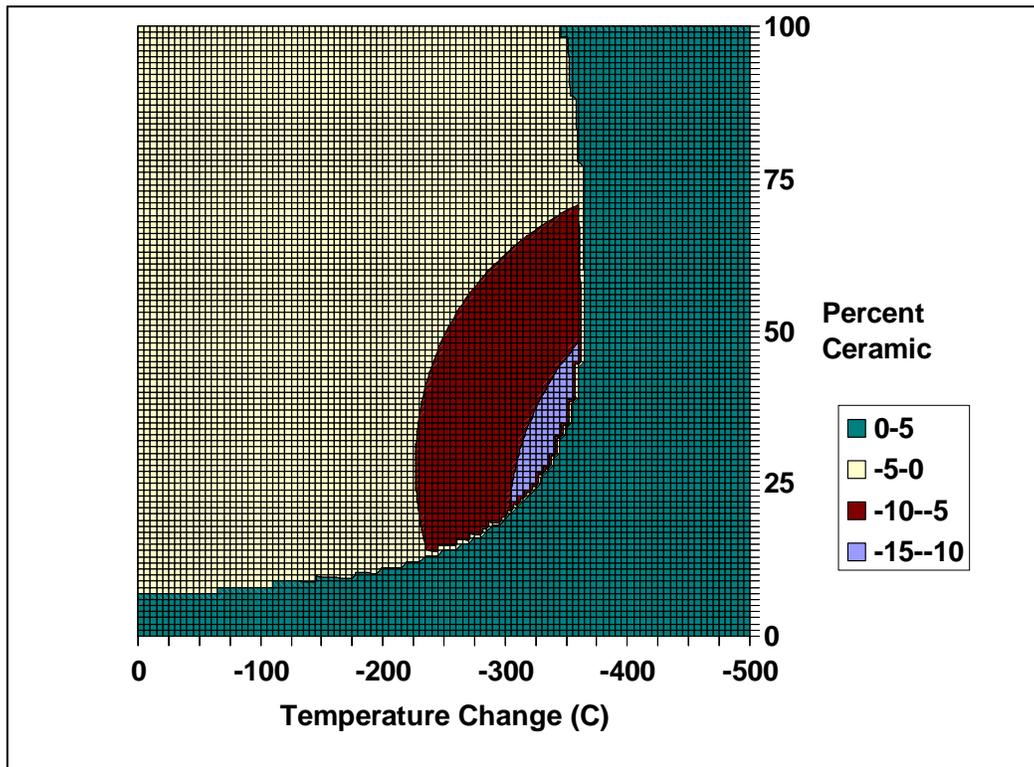


Figure 2. Failure surface for a pressurized tube. The yellow is ceramic failure, green is the sheathing failure, and red and purple are the optimal designs.

one in 1 million chance of failure. Also, the sheathing material can be evaluated for failure as well, allowing for the determination of an optimal design space for the system. For the materials selected here, a Von Mises yield criterion was used to calculate failure in the sheathing layer. When the sheathing failure criterion is met, the value is boosted to a value of one to separate it from the ceramic failures ($\log(P_f)$ is always less than or equal to zero).

For this example, the design space with the optimal chance of success would be to have the wall thickness between 20%–40% ceramic and a ΔT of -300 to -375 °C.

3. Summary

This work derived equations for calculating the effective area and volume and the probability of failure for a ceramic tube subjected to internal and external pressure. The equations have been connected to an elasticity model to calculate the probability of failure for a sheathed ceramic tube. By combining the probability of failure for the ceramic and a failure criterion for the sheath, maps of the optimal design spaces can be generated. A sample calculation demonstrated the ability to model a pressurized, sheathed tube with varying amounts of thermal expansion mismatch.

4. References

1. Weibull, W. A Statistical Theory of the Strength of Materials. *Proceedings of the Royal Swedish Institute of Engineering Research*, 1939, Vol. 151, pp 1–45.
2. ASTM C 1239-00. Standard Practice for Reporting Uniaxial Strength Data and Estimating Weibull Distribution Parameters for Advanced Ceramics. *Annu. Book ASTM Stand.* **2004**.
3. Jadaan, O. M.; Shelleman, D. L.; Conway, J. C., Jr.; Mecholsky, J. J., Jr.; Tressler, R. E. Prediction of the Strength of Ceramic Tubular Components: Part I—Analysis. *Journal of Testing and Evaluation* **1991**, 19 (3), 181–191.
4. Timoshenko, S. P.; Goodier, J. N. Two-Dimensional Problems in Polar Coordinates. In *Theory of Elasticity*; 3rd ed.; McGraw-Hill: New York, NY, 1987; pp 68–71.
5. Johnson, C. A. *Fracture Statistics in Design and Application*; Report No. 79CRD212; General Electric: Schenectady, NY, 1979.
6. Herakovich, C. T. Laminated Tubes. In *Mechanics of Fibrous Composites*; John Wiley & Sons: New York, NY, 1998, 362–401.
7. Rousseau, C. Stresses and Deformations in Angle-Ply Composite Tubes. Master Thesis, Virginia Tech, Blacksburg, VA, 1987.
8. Rousseau, C.; Hyer, M.; Tompkins, S. *Stresses and Deformations in Angle-Ply Composite Tubes*; CCMS-87-04; Virginia Tech: Blacksburg, VA, 1987.
9. Eduljee, R. F.; Gillespie, J. W. Elastic Response of Post- and In Situ Consolidated Laminated Cylinders. *Composites: Part A* **1996**, 27A, 437–446.
10. Carter, R. Model Development for Parametric Design of Pressurized Ceramic Tubing. *Ceramic Engineering & Science Proceedings* **2003**, 24 (4), 477–482.

NO. OF
COPIES ORGANIZATION

1 DEFENSE TECHNICAL
(PDF INFORMATION CTR
ONLY) DTIC OCA
8725 JOHN J KINGMAN RD
STE 0944
FORT BELVOIR VA 22060-6218

1 US ARMY RSRCH DEV &
ENGRG CMD
SYSTEMS OF SYSTEMS
INTEGRATION
AMSRD SS T
6000 6TH ST STE 100
FORT BELVOIR VA 22060-5608

1 INST FOR ADVNCD TCHNLGY
THE UNIV OF TEXAS
AT AUSTIN
3925 W BRAKER LN
AUSTIN TX 78759-5316

1 DIRECTOR
US ARMY RESEARCH LAB
IMNE ALC IMS
2800 POWDER MILL RD
ADELPHI MD 20783-1197

3 DIRECTOR
US ARMY RESEARCH LAB
AMSRD ARL CI OK TL
2800 POWDER MILL RD
ADELPHI MD 20783-1197

3 DIRECTOR
US ARMY RESEARCH LAB
AMSRD ARL CS IS T
2800 POWDER MILL RD
ADELPHI MD 20783-1197

ABERDEEN PROVING GROUND

1 DIR USARL
AMSRD ARL CI OK TP (BLDG 4600)

NO. OF COPIES ORGANIZATION

NO. OF COPIES ORGANIZATION

1 DIRECTOR
US ARMY RESEARCH LAB
AMSRD ARL SE DE
R ATKINSON
2800 POWDER MILL RD
ADELPHI MD 20783-1197

13 COMMANDER
US ARMY ARDEC
AMSTA AR CCH A
F ALTAMURA
M NICOLICH
M PALATHINGUL
D VO
R HOWELL
A VELLA
M YOUNG
L MANOLE
S MUSALLI
R CARR
M LUCIANO
E LOGSDEN
T LOUZEIRO
PICATINNY ARSENAL NJ
07806-5000

5 DIRECTOR
US ARMY RESEARCH LAB
AMSRD ARL WM MB
A ABRAHAMIAN
M BERMAN
M CHOWDHURY
T LI
E SZYMANSKI
2800 POWDER MILL RD
ADELPHI MD 20783-1197

1 COMMANDER
US ARMY MATERIEL CMD
AMXMI INT
5001 EISENHOWER AVE
ALEXANDRIA VA 22333-0001

1 COMMANDER
US ARMY ARDEC
AMSTA AR CCH P
J LUTZ
PICATINNY ARSENAL NJ
07806-5000

2 PM MAS
SFAE AMO MAS MC
PICATINNY ARSENAL NJ
07806-5000

1 COMMANDER
US ARMY ARDEC
AMSTA AR FSF T
C LIVECCHIA
PICATINNY ARSENAL NJ
07806-5000

1 COMMANDER
US ARMY ARDEC
AMSTA AR CC
COL JENKER
PICATINNY ARSENAL NJ
07806-5000

1 COMMANDER
US ARMY ARDEC
AMSTA ASF
PICATINNY ARSENAL NJ
07806-5000

1 COMMANDER
US ARMY ARDEC
AMSTA AR FSE
PICATINNY ARSENAL NJ
07806-5000

1 COMMANDER
US ARMY ARDEC
AMSTA AR QAC T C
J PAGE
PICATINNY ARSENAL NJ
07806-5000

1 COMMANDER
US ARMY ARDEC
AMSTA AR TD
PICATINNY ARSENAL NJ
07806-5000

1 COMMANDER
US ARMY ARDEC
AMSTA AR M
D DEMELLA
PICATINNY ARSENAL NJ
07806-5000

<u>NO. OF COPIES</u>	<u>ORGANIZATION</u>	<u>NO. OF COPIES</u>	<u>ORGANIZATION</u>
3	COMMANDER US ARMY ARDEC AMSTA AR FSA A WARNASH B MACHAK M CHIEFA PICATINNY ARSENAL NJ 07806-5000	1	PM ARMS SFAE GCSS ARMS BLDG 171 PICATINNY ARSENAL NJ 07806-5000
2	COMMANDER US ARMY ARDEC AMSTA AR FSP G M SCHIKSNIS D CARLUCCI PICATINNY ARSENAL NJ 07806-5000	1	COMMANDER US ARMY ARDEC AMSTA AR WEA J BRESCIA PICATINNY ARSENAL NJ 07806-5000
2	COMMANDER US ARMY ARDEC AMSTA AR CCH C H CHANIN S CHICO PICATINNY ARSENAL NJ 07806-5000	1	PM MAS SFAE AMO MAS PICATINNY ARSENAL NJ 07806-5000
1	COMMANDER US ARMY ARDEC AMSTA AR QAC T D RIGIOLIOSO PICATINNY ARSENAL NJ 07806-5000	1	PM MAS SFAE AMO MAS PS PICATINNY ARSENAL NJ 07806-5000
1	US ARMY ARDEC INTELLIGENCE SPECIALIST AMSTA AR WEL F M GUERRIERE PICATINNY ARSENAL NJ 07806-5000	2	PM MAS SFAE AMO MAS LC PICATINNY ARSENAL NJ 07806-5000
9	COMMANDER US ARMY ARDEC AMSTA AR CCH B P DONADIA F DONLON P VALENTI C KNUTSON G EUSTICE K HENRY J MCNABOC R SAYER F CHANG PICATINNY ARSENAL NJ 07806-5000	1	PM MAS SFAE AMO MAS PS PICATINNY ARSENAL NJ 07806-5000
		1	COMMANDER US ARMY TACOM PM COMBAT SYSTEMS SFAE GCS CS 6501 ELEVEN MILE RD WARREN MI 48397-5000
		1	COMMANDER US ARMY TACOM AMSTA SF WARREN MI 48397-5000
		1	DIRECTOR AIR FORCE RESEARCH LAB MLLMD D MIRACLE 2230 TENTH ST WRIGHT PATTERSON AFB OH 45433-7817

NO. OF
COPIES ORGANIZATION

1 OFC OF NAVAL RESEARCH
J CHRISTODOULOU
ONR CODE 332
800 N QUINCY ST
ARLINGTON VA 22217-5600

1 COMMANDER
US ARMY TACOM
PM SURVIVABLE SYSTEMS
SFAE GCSS W GSI H
M RYZYI
6501 ELEVEN MILE RD
WARREN MI 48397-5000

1 COMMANDER
US ARMY TACOM
CHIEF ABRAMS TESTING
SFAE GCSS W AB QT
T KRASKIEWICZ
6501 ELEVEN MILE RD
WARREN MI 48397-5000

1 COMMANDER
WATERVLIET ARSENAL
SMCWV QAE Q
B VANINA
BLDG 44
WATERVLIET NY 12189-4050

2 HQ IOC TANK
AMMUNITION TEAM
AMSIO SMT
R CRAWFORD
W HARRIS
ROCK ISLAND IL 61299-6000

2 COMMANDER
US ARMY AMCOM
AVIATION APPLIED TECH DIR
J SCHUCK
FORT EUSTIS VA 23604-5577

1 NSWC
DAHLGREN DIV CODE G06
DAHLGREN VA 22448

2 US ARMY CORPS OF ENGR
CERD C
T LIU
CEW ET
T TAN
20 MASSACHUSETTS AVE NW
WASHINGTON DC 20314

NO. OF
COPIES ORGANIZATION

1 US ARMY COLD REGIONS
RSCH & ENGRNG LAB
P DUTTA
72 LYME RD
HANOVER NH 03755

13 COMMANDER
US ARMY TACOM
AMSTA TR R
R MCCLELLAND
D THOMAS
J BENNETT
D HANSEN
AMSTA JSK
S GOODMAN
J FLORENCE
D TEMPLETON
A SCHUMACHER
AMSTA TR D
D OSTBERG
L HINOJOSA
B RAJU
AMSTA CS SF
H HUTCHINSON
F SCHWARZ
WARREN MI 48397-5000

14 BENET LABS
AMSTA AR CCB
R FISCELLA
M SOJA
E KATHE
M SCAVULO
G SPENCER
P WHEELER
S KRUPSKI
J VASILAKIS
G FRIAR
R HASENBEIN
AMSTA CCB R
S SOPOK
E HYLAND
D CRAYON
R DILLON
WATERVLIET NY 12189-4050

1 USA SBCCOM PM SOLDIER SPT
AMSSB PM RSS A
J CONNORS
KANSAS ST
NATICK MA 01760-5057

<u>NO. OF COPIES</u>	<u>ORGANIZATION</u>	<u>NO. OF COPIES</u>	<u>ORGANIZATION</u>
1	NSWC TECH LIBRARY CODE B60 17320 DAHLGREN RD DAHLGREN VA 22448	1	NAVAL SEA SYSTEMS CMD D LIESE 1333 ISAAC HULL AVE SE 1100 WASHINGTON DC 20376-1100
2	USA SBCCOM MATERIAL SCIENCE TEAM AMSSB RSS J HERBERT M SENNETT KANSAS ST NATICK MA 01760-5057	7	US ARMY SBCCOM SOLDIER SYSTEMS CENTER BALLISTICS TEAM J WARD W ZUKAS P CUNNIFF J SONG MARINE CORPS TEAM J MACKIEWICZ AMSSB RCP SS W NYKVIST S BEAUDOIN KANSAS ST NATICK MA 01760-5019
2	OFC OF NAVAL RESEARCH D SIEGEL CODE 315 J KELLY 800 N QUINCY ST ARLINGTON VA 22217-5560		
1	NSWC CRANE DIVISION M JOHNSON CODE 20H4 LOUISVILLE KY 40214-5245	7	US ARMY RESEARCH OFC A CROWSON H EVERITT J PRATER G ANDERSON D STEPP D KISEROW J CHANG PO BOX 12211 RESEARCH TRIANGLE PARK NC 27709-2211
2	NSWC U SORATHIA C WILLIAMS CODE 6551 9500 MACARTHUR BLVD WEST BETHESDA MD 20817		
2	COMMANDER NSWC CARDEROCK DIVISION R PETERSON CODE 2020 M CRITCHFIELD CODE 1730 BETHESDA MD 20084	1	AFRL MLBC 2941 P ST RM 136 WRIGHT PATTERSON AFB OH 45433-7750
8	DIRECTOR US ARMY NGIC D LEITER MS 404 M HOLTUS MS 301 M WOLFE MS 307 S MINGLEDORF MS 504 J GASTON MS 301 W GSTATTENBAUER MS 304 R WARNER MS 305 J CRIDER MS 306 2055 BOULDERS RD CHARLOTTESVILLE VA 22911-8318	1	DIRECTOR LOS ALAMOS NATL LAB F L ADDESSIO T 3 MS 5000 PO BOX 1633 LOS ALAMOS NM 87545
		8	NSWC J FRANCIS CODE G30 D WILSON CODE G32 R D COOPER CODE G32 J FRAYSSE CODE G33 E ROWE CODE G33 T DURAN CODE G33 L DE SIMONE CODE G33 R HUBBARD CODE G33 DAHLGREN VA 22448

NO. OF
COPIES ORGANIZATION

1 NSWC
CARDEROCK DIVISION
R CRANE CODE 6553
9500 MACARTHUR BLVD
WEST BETHESDA MD 20817-5700

1 AFRL MLMP
R THOMSON
2977 HOBSON WAY
BLDG 653 RM 215
WRIGHT PATTERSON AFB OH
45433-7739

2 AFRL MLMP
F ABRAMS
J BROWN
2977 HOBSON WAY
BLDG 653 RM 215
WRIGHT PATTERSON AFB OH
45433-7739

5 DIRECTOR
LLNL
R CHRISTENSEN
S DETERESA
F MAGNESS
M FINGER MS 313
M MURPHY L 282
PO BOX 808
LIVERMORE CA 94550

1 AFRL MLS OL
L COULTER
5851 F AVE
BLDG 849 RM AD1A
HILL AFB UT 84056-5713

1 OSD
JOINT CCD TEST FORCE
OSD JCCD
R WILLIAMS
3909 HALLS FERRY RD
VICKSBURG MS 29180-6199

3 DARPA
M VANFOSSEN
S WAX
L CHRISTODOULOU
3701 N FAIRFAX DR
ARLINGTON VA 22203-1714

NO. OF
COPIES ORGANIZATION

1 OAK RIDGE NATL LAB
R M DAVIS
PO BOX 2008
OAK RIDGE TN 37831-6195

1 OAK RIDGE NATL LAB
C EBERLE MS 8048
PO BOX 2008
OAK RIDGE TN 37831

3 DIRECTOR
SANDIA NATL LABS
APPLIED MECHS DEPT
MS 9042
J HANDROCK
Y R KAN
J LAUFFER
PO BOX 969
LIVERMORE CA 94551-0969

1 OAK RIDGE NATL LAB
C D WARREN MS 8039
PO BOX 2008
OAK RIDGE TN 37831

4 NIST
M VANLANDINGHAM MS 8621
J CHIN MS 8621
J MARTIN MS 8621
D DUTHINH MS 8611
100 BUREAU DR
GAITHERSBURG MD 20899

1 HYDROGEOLOGIC INC
SERDP ESTCP SPT OFC
S WALSH
1155 HERNDON PKWY STE 900
HERNDON VA 20170

3 NASA LANGLEY RESEARCH CTR
AMSRD ARL VT
W ELBER MS 266
F BARTLETT JR MS 266
G FARLEY MS 266
HAMPTON VA 23681-0001

1 FHWA
E MUNLEY
6300 GEORGETOWN PIKE
MCLEAN VA 22101

<u>NO. OF COPIES</u>	<u>ORGANIZATION</u>
1	USDOT FEDERAL RAILROAD M FATEH RDV 31 WASHINGTON DC 20590
3	CYTEC FIBERITE R DUNNE D KOHLI R MAYHEW 1300 REVOLUTION ST HAVRE DE GRACE MD 21078
1	DIRECTOR NGIC IANG TMT 2055 BOULDERS RD CHARLOTTESVILLE VA 22911-8318
2	3TEX CORP A BOGDANOVICH J SINGLETARY 109 MACKENAN DR CARY NC 27511
1	DIRECTOR DEFENSE INTLLGNC AGENCY TA 5 K CRELLING WASHINGTON DC 20310
1	COMPOSITE MATERIALS INC D SHORTT 19105 63 AVE NE PO BOX 25 ARLINGTON WA 98223
1	JPS GLASS L CARTER PO BOX 260 SLATER RD SLATER SC 29683
1	COMPOSITE MATERIALS INC R HOLLAND 11 JEWEL CT ORINDA CA 94563
1	COMPOSITE MATERIALS INC C RILEY 14530 S ANSON AVE SANTA FE SPRINGS CA 90670

<u>NO. OF COPIES</u>	<u>ORGANIZATION</u>
1	SIMULA R HUYETT 10016 S 51ST ST PHOENIX AZ 85044
2	PROTECTION MATERIALS INC M MILLER F CRILLEY 14000 NW 58 CT MIAMI LAKES FL 33014
2	FOSTER MILLER M ROYLANCE W ZUKAS 195 BEAR HILL RD WALTHAM MA 02354-1196
1	ROM DEVELOPMENT CORP R O MEARA 136 SWINEBURNE ROW BRICK MARKET PLACE NEWPORT RI 02840
2	TEXTRON SYSTEMS M TREASURE T FOLTZ 1449 MIDDLESEX ST LOWELL MA 01851
1	O GARA HESS & EISENHARDT M GILLESPIE 9113 LESAINTE DR FAIRFIELD OH 45014
1	MILLIKEN RESEARCH CORP M MACLEOD PO BOX 1926 SPARTANBURG SC 29303
1	CONNEAUGHT INDUSTRIES INC J SANTOS PO BOX 1425 COVENTRY RI 02816
1	ARMTEC DEFENSE PRODUCTS S DYER 85 901 AVE 53 PO BOX 848 COACHELLA CA 92236

NO. OF
COPIES ORGANIZATION

3 PACIFIC NORTHWEST LAB
M SMITH
G VAN ARSDALE
R SHIPPELL
PO BOX 999
RICHLAND WA 99352

1 ALLIANT TECHSYSTEMS INC
4700 NATHAN LN N
PLYMOUTH MN 55442-2512

1 APPLIED COMPOSITES
W GRISCH
333 NORTH SIXTH ST
ST CHARLES IL 60174

1 CUSTOM ANALYTICAL
ENG SYS INC
A ALEXANDER
13000 TENSOR LANE NE
FLINTSTONE MD 21530

1 AAI CORP
DR N B MCNELLIS
PO BOX 126
HUNT VALLEY MD 21030-0126

1 OFC DEPUTY UNDER SEC DEFNS
J THOMPSON
1745 JEFFERSON DAVIS HWY
CRYSTAL SQ 4 STE 501
ARLINGTON VA 22202

3 ALLIANT TECHSYSTEMS INC
J CONDON
E LYNAM
J GERHARD
WV01 16 STATE RT 956
PO BOX 210
ROCKET CENTER WV
26726-0210

1 PROJECTILE TECHNOLOGY INC
515 GILES ST
HAVRE DE GRACE MD 21078

1 PRATT & WHITNEY
C WATSON
400 MAIN ST MS 114 37
EAST HARTFORD CT 06108

NO. OF
COPIES ORGANIZATION

5 NORTHROP GRUMMAN
B IRWIN
K EVANS
D EWART
A SHREKENHAMER
J MCGLYNN
BLDG 160 DEPT 3700
1100 W HOLLYVALE ST
AZUSA CA 91701

1 BRIGS COMPANY
J BACKOFEN
2668 PETERBOROUGH ST
HERNDON VA 22071-2443

1 ZERNOW TECHNICAL SERVICES
L ZERNOW
425 W BONITA AVE STE 208
SAN DIMAS CA 91773

2 GENERAL DYNAMICS OTS
FLINCHBAUGH DIV
K LINDE
T LYNCH
PO BOX 127
RED LION PA 17356

1 GKN WESTLAND AEROSPACE
D OLDS
450 MURDOCK AVE
MERIDEN CT 06450-8324

5 SIKORSKY AIRCRAFT
G JACARUSO
T CARSTENSAN
B KAY
S GARBO MS S330A
J ADELMANN
6900 MAIN ST
PO BOX 9729
STRATFORD CT 06497-9729

1 AEROSPACE CORP
G HAWKINS M4 945
2350 E EL SEGUNDO BLVD
EL SEGUNDO CA 90245

2 CYTEC FIBERITE
M LIN
W WEB
1440 N KRAEMER BLVD
ANAHEIM CA 92806

NO. OF
COPIES ORGANIZATION

2 UDLP
G THOMAS
M MACLEAN
PO BOX 58123
SANTA CLARA CA 95052

2 UDLP
R BRYNSVOLD
P JANKE MS 170
4800 E RIVER RD
MINNEAPOLIS MN 55421-1498

1 LOCKHEED MARTIN
SKUNK WORKS
D FORTNEY
1011 LOCKHEED WAY
PALMDALE CA 93599-2502

1 NORTHRUP GRUMMAN CORP
ELECTRONIC SENSORS
& SYSTEMS DIV
E SCHOCH MS V 16
1745A W NURSERY RD
LINTHICUM MD 21090

1 GDLS DIVISION
D BARTLE
PO BOX 1901
WARREN MI 48090

2 GDLS
D REES
M PASIK
PO BOX 2074
WARREN MI 48090-2074

1 GDLS
MUSKEGON OPER
M SOIMAR
76 GETTY ST
MUSKEGON MI 49442

1 GENERAL DYNAMICS
AMPHIBIOUS SYS
SURVIVABILITY LEAD
G WALKER
991 ANNAPOLIS WAY
WOODBIDGE VA 22191

NO. OF
COPIES ORGANIZATION

6 INST FOR ADVANCED
TECH
H FAIR
I MCNAB
P SULLIVAN
S BLESS
W REINECKE
C PERSAD
3925 W BRAKER LN STE 400
AUSTIN TX 78759-5316

1 ARROW TECH ASSOC
1233 SHELBURNE RD STE D8
SOUTH BURLINGTON VT
05403-7700

1 R EICHELBERGER
CONSULTANT
409 W CATHERINE ST
BEL AIR MD 21014-3613

1 SAIC
G CHRYSSOMALLIS
8500 NORMANDALE LAKE BLVD
SUITE 1610
BLOOMINGTON MN 55437-3828

1 UCLA MANE DEPT ENGR IV
H T HAHN
LOS ANGELES CA 90024-1597

1 UMASS LOWELL
PLASTICS DEPT
N SCHOTT
1 UNIVERSITY AVE
LOWELL MA 01854

1 IIT RESEARCH CTR
D ROSE
201 MILL ST
ROME NY 13440-6916

1 GA TECH RESEARCH INST
GA INST OF TCHNLGY
P FRIEDERICH
ATLANTA GA 30392

1 MICHIGAN ST UNIV
MSM DEPT
R AVERILL
3515 EB
EAST LANSING MI 48824-1226

<u>NO. OF COPIES</u>	<u>ORGANIZATION</u>
1	PENN STATE UNIV R S ENGEL 245 HAMMOND BLDG UNIVERSITY PARK PA 16801
1	PENN STATE UNIV C BAKIS 212 EARTH ENGR SCIENCES BLDG UNIVERSITY PARK PA 16802
1	PURDUE UNIV SCHOOL OF AERO & ASTRO C T SUN W LAFAYETTE IN 47907-1282
1	UNIV OF MAINE ADV STR & COMP LAB R LOPEZ ANIDO 5793 AEWB BLDG ORONO ME 04469-5793
1	JOHNS HOPKINS UNIV APPLIED PHYSICS LAB P WIENHOLD 11100 JOHNS HOPKINS RD LAUREL MD 20723-6099
1	UNIV OF DAYTON J M WHITNEY COLLEGE PARK AVE DAYTON OH 45469-0240
5	UNIV OF DELAWARE CTR FOR COMPOSITE MTRLS J GILLESPIE M SANTARE S YARLAGADDA S ADVANI D HEIDER 201 SPENCER LAB NEWARK DE 19716
1	DEPT OF MTRLS SCIENCE & ENGRG UNIV OF ILLINOIS AT URBANA CHAMPAIGN J ECONOMY 1304 W GREEN ST 115B URBANA IL 61801

<u>NO. OF COPIES</u>	<u>ORGANIZATION</u>
1	MISSISSIPPI STATE UNIV DEPT OF AEROSPACE ENGRG A J VIZZINI MISSISSIPPI STATE MS 39762
1	DREXEL UNIV A S D WANG 3141 CHESTNUT ST PHILADELPHIA PA 19104
3	UNIV OF TEXAS AT AUSTIN CTR FOR ELECTROMECHANICS J PRICE A WALLS J KITZMILLER 10100 BURNET RD AUSTIN TX 78758-4497
1	SOUTHWEST RESEARCH INST ENGR & MATL SCIENCES DIV J RIEGEL 6220 CULEBRA RD PO DRAWER 28510 SAN ANTONIO TX 78228-0510
3	DIRECTOR US ARMY RESEARCH LAB AMSRD ARL WM MB A FRYDMAN 2800 POWDER MILL RD ADELPHI MD 20783-1197
<u>ABERDEEN PROVING GROUND</u>	
1	US ARMY ATC CSTE DTC AT AC I W C FRAZER 400 COLLERAN RD APG MD 21005-5059
88	DIR USARL AMSRD ARL CI AMSRD ARL O AP EG M ADAMSON AMSRD ARL SL BA AMSRD ARL SL BB D BELY AMSRD ARL WM J SMITH AMSRD ARL WM B CHIEF T KOGLER AMSRD ARL WM BA D LYON

NO. OF
COPIES ORGANIZATION
 AMSRD ARL WM BC
 J NEWILL
 P PLOSTINS
 AMSRD ARL WM BD
 P CONROY
 B FORCH
 M LEADORE
 C LEVERITT
 R LIEB
 R PESCE-RODRIGUEZ
 B RICE
 A ZIELINSKI
 AMSRD ARL WM BF
 S WILKERSON
 AMSRD ARL WM M
 J MCCAULEY
 S MCKNIGHT
 AMSRD ARL WM MA
 CHIEF
 L GHIORSE
 E WETZEL
 AMSRD ARL WM MB
 J BENDER
 T BOGETTI
 J BROWN
 L BURTON
 R CARTER
 K CHO
 W DE ROSSET
 G DEWING
 R DOWDING
 W DRYSDALE
 R EMERSON
 D GRAY
 D HOPKINS
 R KASTE
 L KECSKES
 M MINNICINO
 B POWERS
 D SNOHA
 J SOUTH
 M STAKER
 J SWAB
 J TZENG
 AMSRD ARL WM MC
 CHIEF
 R BOSSOLI
 E CHIN
 S CORNELISON
 D GRANVILLE
 B HART
 J LASALVIA
 J MONTGOMERY
 F PIERCE

NO. OF
COPIES ORGANIZATION
 E RIGAS
 W SPURGEON
 AMSRD ARL WM MD
 B CHEESEMAN
 P DEHMER
 R DOOLEY
 G GAZONAS
 S GHIORSE
 M KLUSEWITZ
 W ROY
 J SANDS
 D SPAGNUOLO
 S WALSH
 S WOLF
 AMSRD ARL WM RP
 J BORNSTEIN
 C SHOEMAKER
 AMSRD ARL WM T
 B BURNS
 AMSRD ARL WM TA
 W BRUCHEY
 M BURKINS
 W GILLICH
 B GOOCH
 T HAVEL
 C HOPPEL
 E HORWATH
 J RUNYEON
 M ZOLTOSKI
 AMSRD ARL WM TB
 P BAKER
 AMSRD ARL WM TC
 R COATES
 AMSRD ARL WM TD
 D DANDEKAR
 M RAFTENBERG
 S SCHOENFELD
 T WEERASOORIYA
 AMSRD ARL WM TE
 CHIEF
 J POWELL

NO. OF
COPIES ORGANIZATION

1 LTD
R MARTIN
MERL
TAMWORTH RD
HERTFORD SG13 7DG
UK

1 CIVIL AVIATION
ADMINSTRATION
T GOTTESMAN
PO BOX 8
BEN GURION INTRNL AIRPORT
LOD 70150
ISRAEL

1 AEROSPATIALE
S ANDRE
A BTE CC RTE MD132
316 ROUTE DE BAYONNE
TOULOUSE 31060
FRANCE

1 DRA FORT HALSTEAD
P N JONES
SEVEN OAKS KENT TN 147BP
UK

1 SWISS FEDERAL ARMAMENTS
WKS
W LANZ
ALLMENDSTRASSE 86
3602 THUN
SWITZERLAND

1 DYNAMEC RESEARCH LAB
AKE PERSSON
BOX 201
SE 151 23 SODERTALJE
SWEDEN

1 ISRAEL INST OF TECHLGY
S BODNER
FACULTY OF MECHANICAL
ENGR
HAIFA 3200
ISRAEL

1 DSTO
WEAPONS SYSTEMS DIVISION
N BURMAN RLLWS
SALISBURY
SOUTH AUSTRALIA 5108
AUSTRALIA

NO. OF
COPIES ORGANIZATION

1 DEF RES ESTABLISHMENT
VALCARTIER
A DUPUIS
2459 BLVD PIE XI NORTH
VALCARTIER QUEBEC
CANADA
PO BOX 8800 COURCELETTE
GOA IRO QUEBEC
CANADA

1 ECOLE POLYTECH
J MANSON
DMX LTC
CH 1015 LAUSANNE
SWITZERLAND

1 TNO DEFENSE SECURITY & SAFETY
R R IJSSELSTEIN
PO BOX 96864
2509 JG THE HAGUE
THE NETHERLANDS

2 FOA NATL DEFENSE RESEARCH
ESTAB
DIR DEPT OF WEAPONS &
PROTECTION
B JANZON
R HOLMLIN
S 172 90 STOCKHOLM
SWEDEN

2 DEFENSE TECH & PROC
AGENCY GROUND
I CREWTER
GENERAL HERZOG HAUS
3602 THUN
SWITZERLAND

1 MINISTRY OF DEFENCE
RAFAEL
ARMAMENT DEVELOPMENT
AUTH
M MAYSELESS
PO BOX 2250
HAIFA 31021
ISRAEL

1 B HIRSCH
TACHKEMONY ST 6
NETAMUA 42611
ISRAEL

NO. OF
COPIES ORGANIZATION

1 DEUTSCHE AEROSPACE AG
 DYNAMICS SYSTEMS
 M HELD
 PO BOX 1340
 D 86523 SCHROBENHAUSEN
 GERMANY