UNSTEADY COMPRESSIBLE LAMINAR
BOUNDARY-LAYER FLOW WITHIN
AN EXPANSION WAVE

Lang-Mann Chang

July 1983

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does not constitute endorsement of any commercial product.
 Numerical solutions are presented for the unsteady compressible laminar boundary-layer flow developing within a centered expansion wave for two wall thermal conditions, namely, isothermal and adiabatic. The solutions are obtained using Howarth's transformation, a similarity transformation via one-parameter groups, and a series expansion. The expansion in terms of a dimensionless similarity variable is carried up to second-order terms and it improves Hall's zero-order solution for the temperature (Continued on back).
distribution in the boundary layer. It is shown that both velocity and temperature boundary layers grow rapidly behind the expansion wavefront but attenuate further downstream. If the wall temperature is held at the undisturbed temperature, the heat transfer occurs from the wall to the expanding gas. The rate of heat transfer increases from the wavefront to approximately $\xi = 0.35$ behind the wave where maximum heat transfer is observed, $\xi$ being a dimensionless distance from the expansion wavefront. The isothermal wall temperature is found to be higher than the adiabatic wall temperature by a factor of $(1 + 0.335\xi)$. 
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I. INTRODUCTION

In the present report, we study the velocity and temperature boundary-layer development within a centered expansion wave moving into a stationary fluid. A centered expansion wave can be generated either in a shock tube or a tube wind tunnel such as designed by Ludwig.1 It is assumed that the fluid has initially a uniform temperature $T_0$ but is separated by a diaphragm in two regions, with different pressures. If the diaphragm is suddenly removed, an expansion wave propagates into the stationary high pressure region. In the expansion region, the temperature of the fluid decreases due to the drop in the pressure. Therefore, if the wall is kept at the initial temperature $T_0$, there will be a net heat transfer from the wall to the fluid. This heat transfer and the boundary-layer development in the expansion region is the subject of the present analysis.

As early as 1859, Riemann2 analyzed the propagation of a plane disturbance of finite amplitude in an unsteady one-dimensional isentropic flow. Subsequently, many other authors studied the shock tube problem. Huber et al.,3 for example, provided analytical expressions for the velocity, pressure, and temperature of the inviscid flow. However, in order to calculate the heat transfer characteristics, the viscous boundary-layer development on the inner wall must be computed. Mirels4 analyzed the boundary-layer flow behind the expansion wave by assuming that the expansion wave is a line of discontinuity separating the undisturbed high pressure region from the rarefied gas region. Cohen5 used a coordinate expansion method to solve the boundary-layer flow within the centered expansion wave with a finite width. Cohen's treatment, which is an improvement over Mirels' solution, is based on the assumption that the wall is isothermal or, equivalently, that the thermal conductivity of the

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wall is infinite. Hall\(^6\) later generalized Cohen's solution by allowing the wall to have a finite thermal conductivity, and gave solutions of heat transfer for some representative values of thermal conductivity. Hall's solution in the case of adiabatic wall predicts a temperature in the boundary layer that is equal to the temperature in the freestream.

The present study provides solutions for the boundary-layer flow corresponding to two wall thermal conditions, namely, an adiabatic condition and an isothermal condition at undisturbed fluid temperature. The solutions are obtained by applying Howarth's transformation and a similarity transformation via one-parameter groups which reduces the number of independent variables from three to two and then a series expansion. The first three terms of the series are used to obtain a solution of second-order accuracy.\(^7\)–\(^9\)

II. PROBLEM FORMULATION

The boundary-layer flow in the region of the expansion wave as shown in Figure 1 is considered to be two-dimensional, unsteady, compressible, and laminar. The fluid is ideal gas. External to the boundary layer, the flow is assumed to be unsteady, one-dimensional and inviscid. The freestream flow is assumed to be known and it is used as the matching condition for the boundary-layer solution.

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We assume that the undisturbed region is kept at a temperature $T_o$. Then the wavefront of the expansion wave propagates at the sound speed of $U_r = (\gamma RT_o)^{1/2}$. The velocity, temperature, and pressure in the expansion region outside the boundary layer are given by Huber et al. in terms of a similarity variable $\xi$ as follows:

**Velocity**

$$u_i(t,x) = \frac{2}{\gamma + 1} \xi = F(\xi)$$

**Temperature**

$$\theta_i(t,x) = \frac{1}{\gamma} \left(1 - \frac{\gamma - 1}{\gamma + 1} \xi\right)^2 = G(\xi)$$

**Pressure**

$$p_i(t,x) = (1 - \frac{\gamma - 1}{\gamma + 1} \xi)^{2\gamma/(\gamma-1)} = H(\xi)$$

where the similarity variable $\xi$, derived from the similarity transformation via one-parameter groups is

$$\xi = 1 + \left(\frac{x}{t}\right) , \quad x = \frac{\bar{x}}{L}, \quad t = \frac{\bar{t}}{L/U_r}$$
and $L$ is a reference length. As shown in Figure 1, $\xi = 0$ denotes the expansion wavefront and $\xi = 1$ is located at the origin of the wave. $F(\xi)$, $G(\xi)$, and $H(\xi)$ are, respectively, called the velocity function, temperature function, and pressure function of the inviscid flow. The velocity $u_1$ is the inviscid velocity normalized by $U_r$, the temperature $\theta_1$ by the reference temperature $T_r = U_r^2/R$, and the pressure $p_1$ by $\rho_o U_r^2$. $P_o$ is the density of the gas at rest ahead of the expansion wave.

In the boundary layer we assume that the thermal conductivity $k$ and specific heat $C_p$ are constants and that the viscosity $\mu$ is proportional to the temperature. The viscosity $\bar{\mu}$ can be made dimensionless as

$$u = \bar{\mu}/\mu_r = T/T_r$$

where the reference viscosity $\mu_r$ is evaluated at the reference temperature $T_r$.

We also define the following dimensionless variables and parameters:

$$y' = \text{Re}^{1/2} Y/L, \quad u = U/U_r,$$

$$v = \text{Re}^{1/2} V/U_r, \quad p = P/(\rho_o U_r^2), \quad \theta = T/T_r, \quad \rho = \bar{\rho}/\rho_o,$$

$$\text{Re} = U_r L \rho_o/\mu_r, \quad \text{Pr} = C_p \mu_r/k, \quad T_r = U_r^2/R.$$

The governing equations for the two-dimensional, unsteady, compressible boundary-layer flow are, in terms of these variables, 

**Continuity**

$$\frac{3p}{3t} + \frac{3}{3x} (\rho u) + \frac{3}{3y} (\rho v) = 0$$

**Momentum**

$$\rho (\frac{3u}{3t} + u \frac{3u}{3x} + v \frac{3u}{3y}) = - \frac{dp}{dx} + \frac{3}{3y} (\mu \frac{3u}{3y})$$
Energy

\[ \rho \left( \frac{\partial \theta}{\partial t} + u \frac{\partial \theta}{\partial x} + v \frac{\partial \theta}{\partial y} \right) = \frac{1}{Pr} \frac{\partial}{\partial y} \left( u \frac{\partial \theta}{\partial y} \right) \]

(9)

\[ + \frac{1}{\gamma} \left[ \frac{\partial p}{\partial t} + u \frac{\partial p}{\partial x} + \theta \left( \frac{\partial u}{\partial y} \right)^2 \right] \]

State

\[ p = \rho \theta \]

(10)

The pressure \( p \) in Eq. (8) is the same as the \( p_1 \) in Eq. (3). By introducing Howarth's transformation

\[ y = \int^y \rho \, dy' \]

(11)

a streamfunction \( \psi \), which satisfies the continuity equation, can be defined as

\[ u = \frac{\partial \psi}{\partial y}, \quad v = -\frac{1}{\rho} \left( \frac{\partial \psi}{\partial x} + \frac{\partial \psi}{\partial y} \frac{\partial y}{\partial x} + \frac{\partial y}{\partial t} \right) \]

(12)

Substituting Eqs. (11) and (12) into Eqs. (8) and (9), we have

\[ \frac{\partial^2 \psi}{\partial t \partial y} + \frac{\partial \psi}{\partial y} \frac{\partial^2 \psi}{\partial x \partial y} - \frac{\partial \psi}{\partial x} \frac{\partial^2 \psi}{\partial y^2} = -\frac{\partial p}{\partial x} + \frac{\partial^3 \psi}{\partial y^3} \]

(13)

and

\[ \frac{\partial \theta}{\partial t} + \frac{\partial \psi}{\partial y} \frac{\partial \theta}{\partial x} - \frac{\partial \psi}{\partial x} \frac{\partial \theta}{\partial y} = \frac{p}{Pr} \frac{\partial^2 \theta}{\partial y^2} + \frac{1}{\gamma} \left( \frac{\partial^2 \psi}{\partial y^2} \right) \]

(14)

In addition, we have the following boundary conditions for \( \psi \) and \( \theta \):
At the wall, $y = 0$

$$u = \frac{\partial \psi}{\partial y} = 0 \quad \text{(no slip condition)}$$  \hspace{1cm} (15)

$$v = -\frac{1}{\rho} \frac{\partial \psi}{\partial x} = 0 \quad \text{(impermeable condition)}$$

$$\theta = \theta_w = \frac{1}{y} \quad \text{(isothermal wall at $T_o$)}$$  \hspace{1cm} (16)

or

$$\frac{\partial \theta}{\partial y} = 0 \quad \text{(adiabatic wall)}$$  \hspace{1cm} (17)

At the outer edge of the boundary layer, $y \to \infty$

$$u = \frac{\partial \psi}{\partial y} = u_i(t,x)$$  \hspace{1cm} (18)

$$\theta = \theta_i(t,x)$$  \hspace{1cm} (19)

III. METHOD OF SOLUTION

We obtain a solution of Eqs. (13) and (14) through a similarity transformation via one-parameter groups and a subsequent series expansion of the functions $\psi$ and $\theta$. The similarity transformation reduces the three independent variables $(t,x,y)$ to two similarity variables $(\xi, \eta)$ which are

$$\xi = 1 + \frac{x}{t} \quad \text{and} \quad \eta = \frac{y}{t^{1/2}}$$  \hspace{1cm} (20)

A detailed derivation of these two similarity variables is given by Chang.\(^\text{10}\)

The similarity transformation also transforms the dependent variables, $\psi$ and $\theta$ in Eqs. (13) and (14) into

Substitution of Eqs. (3) and (20-22) into Eqs. (13) and (14) results in

\[ (1-\xi)\xi - \frac{1}{2} \eta \xi + \xi \xi - \xi \xi \eta = - \frac{g}{H} + \frac{Hf}{H} \eta \eta \eta \]  

(23)

and

\[ (1-\xi)\xi - \frac{1}{2} \eta \xi + \xi \xi - \xi \xi \eta = \frac{1}{Pr} \frac{Hg}{H} \eta \eta \eta \]  

\[ + \frac{1}{\gamma} \left[ \frac{(1-\xi)H' + \xi \xi H' + \frac{H^2}{g} \xi \xi \xi}{H} \right] \]  

where \( H' \) is the derivative of \( H(\xi) \) with respect to \( \xi \). A repeated application of the transformation technique shows that no new similarity variables that combine the variables \( \xi \) and \( \eta \) exist. However, we can define a locally similar variable as

\[ z = \eta / \xi^{1/2} \]  

(25)

Using the new variable \( z \), a fast-converging series solution to the problem may be constructed as follows

\[ f = \xi^{1/2}F(\xi) = \sum_{n=0}^{\infty} \xi^n f_n(z) \]  

(26)

\[ u = \frac{3\psi}{\partial y} = \frac{3f}{\partial \eta} = F(\xi) \sum_{n=0}^{\infty} \xi^n f'_n(z) \]  

(27)

\[ \theta = g(\xi, \eta) = \frac{1}{\gamma} + \left[ G(\xi) - \frac{1}{\gamma} \right] \sum_{n=0}^{\infty} \xi^n g_n(z) \]  

(28)

Substituting the above expressions and their derivatives into Eqs. (23) and (24), we have, after grouping the terms with the same power of \( \xi \), the following system of ordinary differential equations.
Zero-order equations (n=0)

\[ f''_0 + \frac{1}{2}zf''_0 - f'_0 = -1 \]  \hspace{1cm} (29)

\[ \frac{1}{Pr}g''_0 + (1/2)zg'_0 - g_0 = -1 \]  \hspace{1cm} (30)

First-order equations (n=1)

\[ f''_1 + \frac{1}{2}zf''_1 - 2f'_1 = \frac{2\gamma}{\gamma+1} f''_0 - \frac{3}{\gamma+1} f_0 f'_0 - f'_0 \]  \hspace{1cm} (31)

\[ + \frac{2}{\gamma+1} (f'_0)^2 + 2 \frac{\gamma-1}{\gamma+1} g_0 - \frac{\gamma-1}{\gamma+1} \]

\[ \frac{1}{Pr} g''_1 + \frac{1}{2}zg''_1 - 2g_1 = \frac{1}{Pr} \frac{5\gamma-1}{2(\gamma+1)} g''_0 + (\frac{\gamma-1}{4} \frac{\gamma+1} {g'_0} g''_0 \]

\[ - (\frac{2}{\gamma+1}) g_0 - (\frac{3}{\gamma+1}) g'_0 f'_0 + (\frac{2}{\gamma+1}) f''_0 \]  \hspace{1cm} (32)

\[ + (\frac{2}{\gamma+1}) f'_0 + (\frac{2}{\gamma+1}) f'_0 g_0 + \frac{2}{\gamma+1} \]

Second-order equations (n=2)

\[ f''_2 + (1/2)zf''_2 - 3f'_2 = F_*(z) \]  \hspace{1cm} (33)

\[ \frac{1}{Pr}g''_2 + (1/2)zg'_2 - 3g_2 = G_*(z) \]  \hspace{1cm} (34)

The expressions \( F_*(z) \) and \( G_*(z) \) are given in the Appendix. The boundary conditions, Eqs. (15) - (19), become:

at the wall, \( z = 0 \), for \( n > 0 \)

\[ f_n(0) = 0, \quad f'_n(0) = 0 \]
\( g_n(0) = 0 \) for isothermal wall \hspace{1cm} (35)

or

\( g_n'(0) = 0 \) for adiabatic wall

at the outer edge of the boundary layer, \( z = \infty \)

\( f_0'(\infty) = 1, \quad f_n'(\infty) = 0 \) for \( n \geq 1 \)

\( g_0(\infty) = 1, \quad g_n(\infty) = 0 \) for \( n \geq 1 \) \hspace{1cm} (36)

\( g_n'(\infty) = 0 \) for \( n > 0 \)

IV. NUMERICAL RESULTS AND DISCUSSION

The preceding three sets of ordinary differential equations were solved numerically. \(^{10}\) With \( Pr = 0.72 \) and \( \gamma = 1.4 \) the numerical results of the velocity functions \( f_0', f_1', \) and \( f_2' \), and the temperature functions \( g_0, g_1, \) and \( g_2 \) for both isothermal and adiabatic walls are tabulated in Tables 1 and 2 and plotted in Figures 2 and 3. Since there is no temperature term involved in the zero-order momentum equation, Eq. (29), the function \( f_0' \) is independent of the temperature condition at the wall. Therefore, as shown in Figure 2, the velocity function \( f_0' \) is the same for the two wall conditions. However, the function \( f_1' \) and \( f_2' \) are certainly affected by the wall temperature. For the case of isothermal wall the present results are identical to Cohen's solutions. \(^{10}\)

Once the zero-order, first-order, and second-order functions are computed, the solutions to the problem in a three-term series can be readily constructed.
**Table 1. Velocity Functions**

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**Table 2. Temperature Functions**

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A. Velocity Profiles

From Eq. (27), the velocity in the boundary layer is simply

$$u = F\left(f_0' + \xi f_1' + \xi^2 f_2'\right)$$

(37)

or

$$u/u_1 = f_0' + \xi f_1' + \xi^2 f_2'$$

(38)

for $F = u_1$ given in Eq. (1). Figures 4 and 5 show $u/u_1$ vs $z$, respectively, for isothermal and adiabatic walls. One observes that at a given $z$, $u/u_1$ increases
with $\xi$. This is because, as shown in Eq. (1), the inviscid velocity $u_i$ relative to the wavefront appears to accelerate in the positive $\xi$ direction. In other words, the fluid is expanding from the high pressure to the lower pressure region and the further the downstream distance is from the wavefront the higher the fluid velocity becomes. As a result, even though the viscosity diffusion in the boundary layer tends to increase the boundary growth the acceleration may suppress some growth of the boundary layer. If the boundary-layer thickness is taken to be approximately $z_\delta = 4$ from Figures 4 and 5, Eqs. (25) and (20) give

$$z_\delta = y_\delta / x + t = 4$$

(39)

Thus, at a given instance, the boundary-layer thickness $y_\delta$ behind the wavefront may grow like $4 / x + t$. The larger the time, the slower the growth of the boundary-layer thickness will be with respect to $x$. In other words, the boundary layer grows rapidly immediately behind the expansion wavefront and flattens out further downstream from the wavefront.

![Figure 4. Velocity Profiles for an Isothermal Wall at $\xi = 0, 0.1, 0.2, \text{and} 0.3$](image)

![Figure 5. Velocity Profiles for an Adiabatic Wall at $\xi = 0, 0.1, 0.2, \text{and} 0.3$](image)
A careful comparison between the profiles shown in Figures 4 and 5 reveals that at a given z the velocity is slightly higher for the isothermal wall due to heat addition from the wall to the fluid.

B. Temperature Profiles

The temperature in the boundary layer is

\[ \theta = \frac{1}{\gamma} + (G - \frac{1}{\gamma}) \left( g_o + \xi g_1 + \xi^2 g_2 \right) \]

or

\[ \frac{\theta}{\theta_1} = \left[ \frac{1}{\gamma} + (G - \frac{1}{\gamma}) \left( g_o + \xi g_1 + \xi^2 g_2 \right) \right] / G \] (40)

The solutions are plotted in Figure 6. The temperature in the boundary layer is seen to be higher than that in the freestream and the value of \( \theta/\theta_1 \) increases with \( \xi \) as a result of an increasing viscous heating in the boundary layer. Figure 6 shows that at \( \xi = 0.3 \) the wall temperature is higher than the local inviscid temperature by approximately 10 percent.

Figure 6. Temperature Profiles at \( \xi = 0, 0.1, 0.2, \) and 0.3

Figure 6 also shows that because of heat addition to the fluid from the wall, the surface temperature for the isothermal wall is higher than that for the adiabatic wall by 3.34 percent at \( \xi = 0.1 \), 6.6 percent at \( \xi = 0.2 \), and 10.9 percent at \( \xi = 0.3 \); giving the temperature ratio \( T_w \) (isothermal) to \( T_w \) (adiabatic) of approximately \( (1 + 0.335\xi) \). Hall considered the case with a finite wall thermal conductivity. He found that the temperature in the
boundary layer on an adiabatic wall is the same as the temperature in the inviscid freestream. This result is equivalent to the zero-order solution (i.e., $\xi = 0$) of Eq. (40).

C. **Heat Transfer**

For an isothermal wall the local heat transfer from the wall to the fluid is

$$ Q_w = - k \left. \frac{\partial T}{\partial y} \right|_{y=0} $$

(41)

Written in dimensionless form, it becomes

$$ q_w = - \left. \frac{\partial \Theta}{\partial y'} \right|_{y'=0} $$

(42)

Transformed into $(\xi,z)$ coordinate system, it is

$$ q_w = -t^{-1/2} \gamma \xi^{-1/2} \left( g_0' + \frac{1}{\gamma} g_1' + \xi^2 g_2' \right) $$

(43)

The values of $g_0'$, $g_1'$, and $g_2'$ are, respectively, 0.9575, 0.15074, and -0.25445. Figure 7 and Table 3 show that at a given instant the heat transfer increases rapidly behind the wavefront until it approaches a maximum, and then likely decreases. This can be explained as a result of an interplay among the expansion of the freestream, the growing boundary-layer thickness, and the viscous heating. From Eq. (2) we notice that the freestream temperature decreases with increasing distance from the wavefront. Consequently, heat transfer is from the wall to the fluid. On the other hand, the combined effect of the growing boundary-layer thickness and the viscous heating provides a resistance to the heat transfer further downstream. Equation (43) shows that the heat transfer at a given location decreases with respect to time and is approximately proportional to $t^{-1/2}$. Physically this is reasonable because at a given location the boundary layer grows in time after the expansion wave passes by. The growth of the boundary layer deters the heat transfer.
The coefficient of local heat transfer is determined by Newton's cooling law

\[ Q_w = h(T_w - T_i) \]
or

\[ h = q_w R(\text{Re} \ k)^{-1/2} \left[ \frac{1}{(1/\gamma) - G} \right]^{-1} \]  

(44)

where \( R \) is the gas constant. The function \( G \) is given in Eq. (2).

D. Skin Friction

The friction coefficient on the wall is defined as

\[ C_f = \frac{1}{\rho U^2} \left. \frac{2U}{\partial Y} \right|_{Y=0} \]  

(45)

After coordinate transformation and series expansion, Eq. (45) becomes

\[ C_f = (\text{Re} \ t \xi)^{-1/2} \left[ f''_0(0) + \xi f''_1(0) + \xi^2 f''_2(0) \right] \]  

(46)

The values of \( f''_0, f''_1(0), \) and \( f''_2(0) \) are, respectively, 1.1284, 0.7946, 0.41781 for the isothermal wall and 1.1284, 0.65896, 0.32903 for the adiabatic wall. The plot of \( C_f(\text{Re} \ t \xi)^{-1/2} \) is also given in Figure 7 and Table 3. There is no large difference in skin friction between the isothermal wall and the adiabatic wall because both have very similar velocity profiles as shown in Figures 4 and 5. Specifically, at \( \xi = 0.3 \) the skin friction for the isothermal wall is approximately 5.5 percent higher than that for the adiabatic wall.

V. SUMMARY AND CONCLUSIONS

Solutions were obtained for the unsteady compressible laminar boundary-layer flow that develops within a centered expansion wave for isothermal and adiabatic walls. The solutions were obtained by a method of similarity transformation via one-parameter groups and a power series expansion in terms of a dimensionless distance \( \xi \) from the expansion wavefront. The series expansion includes terms up to second order of \( \xi \), and thus constitutes an improvement over Hall's zero-order solution for the temperature in the boundary layer over an adiabatic wall. The present results show that at a given location the isothermal wall surface temperature is higher than the adiabatic wall surface temperature by a factor of approximately 0.335 \( \xi \).
REFERENCES


APPENDIX A

Function $F^*$ in Eq. (33) and Function $G^*$ in Eq. (34)
Function $F^*(z)$ of Eq. (33)

$$F^*(z) = (\frac{3y^{-1}}{y+1})f_1''' - (\frac{1}{2} \frac{y^{-1}}{y+1})zf_1'' - (\frac{3}{y+1})f_0'f_1'' - (\frac{5}{y+1})f_0''f_1'$$

$$- (\frac{5}{2} + 2 \frac{y^{-1}}{y+1})f_1' + (\frac{6}{y+1})f_0'f_1' - (\frac{5}{y+1})f_0f_1$$

$$- 3y^2 \frac{y}{(y+1)^2} f_0' + 3(\frac{y^{-1}}{(y+1)^2})f_0f_0' + \frac{3}{2} \frac{y^{-1}}{y+1} f_0$$

$$- 2 \frac{y^{-1}}{(y+1)^2} f_0'f_0' + 2 \frac{y^{-1}}{y+1} g_1 - (\frac{y^{-1}}{y+1})^2 g_2$$

Function $G^*(z)$ of Eq. (34)

$$G^*(z) = \frac{1}{Pr} \left( \frac{1}{2} \frac{y^{-1}}{y+1} + \frac{2y}{y+1} \right) g_1'' + \left( \frac{1}{4} \frac{y^{-1}}{y+1} z \right) g_1$$

$$- \frac{3}{y+1} f_0'g_1''' - (2 - \frac{y^{-1}}{2(y+1)})g_1' + \frac{4}{y+1} f_0g_1$$

$$- \frac{1}{Pr} \left( \frac{y(y^{-1})}{(y+1)^2} + \frac{y}{y+1} \right) g_1'' - \frac{5}{y+1} f_1g_0$$

$$+ \left( \frac{3}{2} \frac{y^{-1}}{(y+1)^2} \right) f_0g_0' - (\frac{y^{-1}}{y+1} - (\frac{y^{-1}}{y+1})^2)g_0$$

$$- \frac{4y}{(y+1)^2} f_0'' + \frac{2(y^{-1})}{(y+1)^2} f_0' - \frac{2}{y+1} f_1' + (2 \frac{y^{-1}}{(y+1)^2})f_0'g_0$$

$$+ \frac{2}{y+1} f_1'g_0 + \frac{y^{-1}}{y+1} - (\frac{y^{-1}}{y+1})^2 - \frac{4}{y+1} f_0''f_1$$
NOMENCLATURE

\(C_f\) = coefficient of local skin friction

\(C_p\) = specific heat at constant pressure

\(F\) = inviscid velocity function, Eq. (1)

\(F^*\) = right-hand side of Eq. (33), Appendix A

\(f\) = transformed streamfunction, Eq. (21)

\(f_n\) = transformed streamfunction, Eq. (26), \(n = 1,2,3,\ldots\)

\(G\) = inviscid temperature function, Eq. (2)

\(G^*\) = right-hand side of Eq. (34), Appendix A

\(g\) = transformed temperature function, Eq. (22)

\(g_n\) = transformed temperature function, Eq. (28)

\(H\) = inviscid pressure function, Eq. (3)

\(h\) = heat transfer coefficient

\(k\) = thermal conductivity

\(L\) = reference length

\(n\) = integer, \(n = 0,1,2,\ldots\)

\(p\) = dimensionless pressure

\(\bar{p}\) = dimensional pressure

\(P_i\) = dimensionless freestream pressure

\(Pr\) = Prandtl number

\(Re\) = Reynolds number

\(Q_w\) = dimensional heat flux at the wall

\(q_w\) = dimensionless heat flux at the wall

\(R\) = gas constant
$T$  = dimensional temperature

$T_i$  = dimensional freestream temperature

$T_0$  = dimensional temperature of the undisturbed fluid

$T_r$  = reference temperature $U_r^2/R = \gamma T_0$

$T_w$  = dimensional wall temperature

$t$  = dimensionless time $tU_r/L$

$t^*$  = dimensional time

$t^*$  = dimensionless time, Figure 1

$U$  = $x$-component velocity, dimensional

$U_r$  = reference velocity, speed of the expansion wavefront $\sqrt{\gamma R T_0}$

$u$  = $x$-component velocity, dimensionless, Eq. (6)

$u_i$  = dimensionless freestream velocity

$V$  = $y$-component velocity, dimensional

$v$  = $y$-component velocity, dimensionless, Eq. (6)

$x, y'$  = dimensionless coordinates, Figure 1 and Eq. (6)

$\bar{x}, \bar{y}$  = dimensional coordinates

$y$  = dimensionless, transformed coordinate $y'$, Eq. (11)

$z$  = variable, Eq. (25)

$\gamma$  = ratio of specific heats

$n$  = similarity variable, Eq. (20)

$\theta$  = dimensionless temperature

$\theta_i$  = dimensionless freestream temperature

$\theta_w$  = dimensionless wall temperature

$\mu$  = dimensionless viscosity
- $\mu$ = dimensional viscosity
- $\mu_r$ = reference viscosity
- $\xi$ = similarity variable, Eq. (4)
- $\rho$ = dimensionless density, Eq. (6)
- $\rho$ = dimensional density
- $\rho_r$ = reference density
- $\psi$ = streamfunction, Eq. (12)
- $\gamma$ = outer edge of the boundary layer

Subscripts
- $i$ = inviscid
- $r$ = reference
- $w$ = wall condition
- $0$ = undisturbed
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